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# **Spatial-Spectral Sensor Techniques for Detection of Atmospheric Turbulence**

## **Phase I Final Report**

**Small Business Innovative Research Program**

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**March 3, 2000**

**Final Report by**

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This report addresses the application of spatial-spectral (hyperspectral) imaging from space-borne or multi-platform sensors to detect and characterize atmospheric turbulence. Turbulence may impact performance of optical systems deployed for energy propagation, such as airborne and space-based lasers, and sensors systems used for dim target detection, such as aircraft or cruise missile surveillance and tracking. Techniques developed allow global scientific investigation of regions of atmospheric turbulence and clutter, including the study of cirrus clouds. Potential commercial applications include improved detection and understanding of clear air turbulence to improve aircraft safety. Military applications include improved understanding of the effect of the atmospheric on propagation of laser energy and the detection and tracking of dim targets such as cruise missiles.

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## **PREFACE**

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## 1. INTRODUCTION

This report addresses the problem of utilizing spatial-hyperspectral imaging capabilities of space-borne sensors to detect and characterize regions of atmospheric turbulence and cirrus cloud clutter which may impact employment and/or performance of space based laser and airborne high energy laser weapons systems.

Our concept for worldwide detection, characterization and mapping of atmospheric turbulence and cirrus clouds involves use of satellite-borne (and possibly airborne) spectral and hyperspectral imagers operated in the UV to MWIR spectral range in virtual triangulation geometry. Spectral and hyperspectral imagery allows altitude sounding of atmospheric clutter from turbulence and cirrus clouds. Triangulation geometry allows precise altitude selection by cross correlation of the backscatter signals. The combination of altitude and Fourier-space background spectral discrimination will provide an altitude resolved measurement of atmospheric clutter from clear air turbulence and from cirrus clouds, both of which may affect performance of the SBL (Space Based Laser) and the Airborne Laser (ABL) systems.

Figure 1 illustrates the virtual triangulation concept by forward-looking and backward-looking imaging sensors from a satellite.

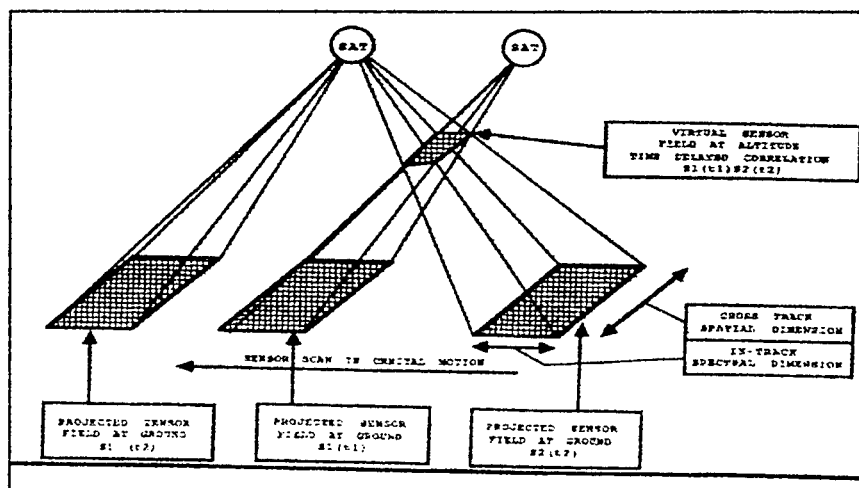


Figure 1 Conceptual Optical Triangulation Geometry to Isolate Atmospheric Turbulence, Characterize Ground-based Clutter, and Detect Weak Targets. The virtual triangulation geometry allows mapping of random atmospheric clutter and CAT vs. altitude by cross correlation of forward-viewing and backward-viewing imaging signals and assists in separating weak airborne targets from the atmospheric background clutter.

We anticipate that the conceptual satellite sensor can be tested on ground-based and airborne platforms prior to finalization of a satellite sensor design. This phase one SBIR project has initiated such conceptual tests but further work is required, especially with respect to airborne tests of the concept versus clear air turbulence (CAT).

Anticipated commercial applications of this research are automated background scene content identification and weak target recognition by small, real-time programmable space-borne sensors and remote sensing of atmospheric structure which leads to a capability for mapping CAT (clear air turbulence) from airborne or space-borne detectors.

## **2. STATEMENT OF PROBLEM AND MEASUREMENT CONCEPT**

### **2.1.1 Technical Problem Summary**

The technical problem addressed by this SBIR program is to devise unique spatial-spectral satellite-based observation techniques to detect weak targets in the presence of background clutter. The weak targets that we address are in order of observability:

- small, low observable airborne vehicles,
- thin atmospheric cirrus clouds,
- atmospheric clear air turbulence, and
- turbulent structure in the troposphere, stratosphere, and lower thermosphere.

This report concentrates on the more difficult “targets”, atmospheric turbulence and clear air turbulence (CAT) layers. The rationale for measurement of the atmospheric quantities is that this class of backgrounds forms the limiting cases of background clutter, which if measurable by a satellite passive sensor ensures that other airborne targets, will be detectable.

### **2.1.2 Technical Approach**

We validated the virtual triangulation – spectral – spatial weak target detection and characterization technique by analytical simulation techniques and by experimental simulation using ground-based spectral-spatial structure sensors and by using the GLO sensors carried on the Space Shuttle<sup>1</sup>.



### **Validation by Simulation**

We used the NSS (Non-stationary Stochastic Structure) Model<sup>2</sup> and LAMSS<sup>3</sup> (Low Altitude Mesoscale Stochastic Structure) models to simulate structure in the mesopause region (~90 km) and in the troposphere (< 20 km) respectively. These simulations were generated using an geometrical driver for the sensor LOS to generate an artificial data stream appropriate to the experimental validation geometries.

### **Validation by Experiment**

Two experiments were conducted to validate the satellite sensor cross-beam concept: a low altitude experiment measured turbulent structure and winds in the boundary layer between about 50 and 300 feet altitude. We intended that the daytime measurements would be coordinated with meteorological tower measurements of winds and turbulence but this experiment could not be consummated under the resources of the project. A high altitude set of measurements was conducted at nighttime used the airglow emission in the 80 to 130 km altitude range to provide the signal. Daytime experiments used the Kestrel AIRCAM sensors. operating in the 450 to 950 nm spectral range. The nighttime high altitude structure experiments used the Fourier Transform Hyperspectral Imager FTHSI and supporting imaging instrumentation from LANL and University of Arizona.

#### **2.1.3 Results Summary**

The Phase I simulations and ground-based experiments validated to a certain extent the ability of a satellite-borne passive atmospheric structure sensor to provide worldwide measurements of atmospheric turbulence and wind structure especially thin cirrus clouds and clear air turbulence. Validation experiment results may be applied to further experiments to determine atmospheric structure phenomenology and clutter maps for the Airborne Laser System (ABL), the Space-based Laser System (SBL), CAT maps for the civilian sector, and detection weak targets through efficient mitigation of of atmospheric background structure-induced clutter.

### 3. CONCEPT VALIDATION

The atmospheric structure sounding concept was validated by two techniques. Simulation of atmospheric structure and synthesis of atmospheric backscatter signals for low altitude structure and atmospheric airglow signals for high altitude structure was carried out for a limited number of cases and scenarios. Experiments designed to simulate possible satellite and/or airborne sensor environments and their respective signals generated by structure were conducted to investigate the utility of Hyperspectral Imagery. Data from the GLO sensor that has been repeatedly flown on the Space Shuttle also contributed to our experimental database.

#### 3.1 VALIDATION BY SIMULATION

##### 3.1.1 Simulation of Winds and Turbulence

Atmospheric turbulence was simulated in two dimensions by the technique demonstrated by Strugala and Sears, et. al. (See reference ). Two different two-dimensional spectra were adopted: a  $k^{-5/3}$  power law approximates the turbulence spectrum for an optically thick medium, and a  $k^{-8/3}$  power law spectrum approximates an optically thin turbulent medium. Both spectra are of the modified Kolmogorov type with a correlation length of 10 cm, about the value expected for the lower atmosphere.

Equation 1 illustrates the functional form of the spectra for a two dimensional scene:

$$P(k_x, k_y) = [\Gamma(p) \Gamma(p) L_{cx} L_{cy}] / \{k_x/L_{cx}\}^2 + \{k_y/L_{cy}\}^2\}^p \quad (1)$$

Where,

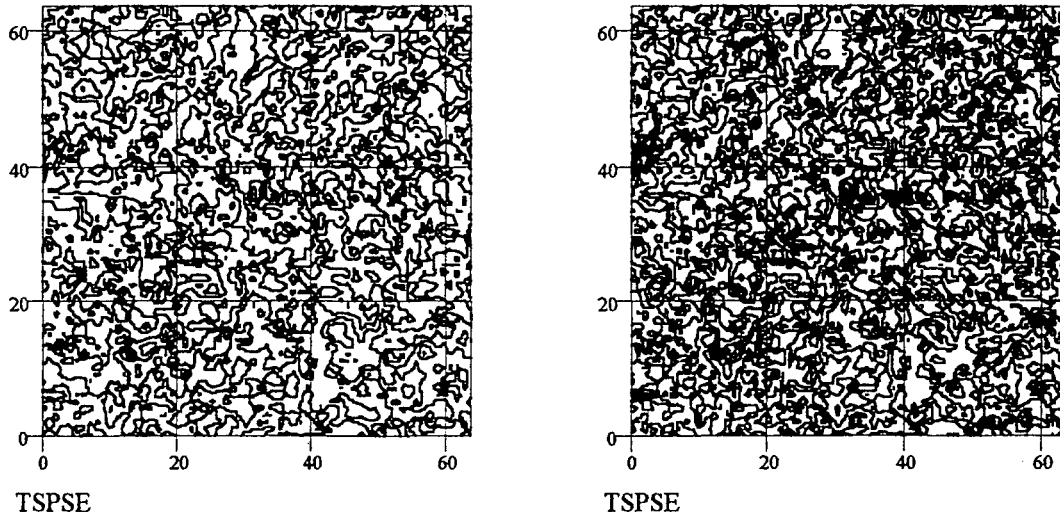
$p = 5/6$  or  $8/6$  for  $k^{-5/3}$  or  $k^{-8/3}$  spectral forms respectively,

$n = (p-1)/p$ ,

$k_{x,y}$  is the spatial frequency in either x or y dimension

$\Gamma(n)$  is the gamma function with argument determined by the spectral index

The background spatial scenes were implemented using the method of Strugala and Sears wherein a random Gaussian number field is created, then convolved with the desired 2-D spectral function (equation 1) to achieve a correlated random background intensity field. Figure 2 illustrates the two dimensional scenes generated by the simulation routine. Here, we used the same random number set in two dimensions to generate the arrays and vary only the power spectral density slope. A 64x64 pixel sample was cut from each 256x256 scene to show that the higher slope ( $k^{-8/3}$ ) has much more high frequency structure as indicated by the density of the contour lines. Each picture was digitized to a six-level contour interval.



**Figure 2 Contour maps of 2-D scenes generated from the same Gaussian random number field, but with spectral indices of  $k^{-5/3}$  (left) and  $k^{-8/3}$  (right) which physically corresponds to Kolmogorov turbulence in an optical thick and optically thin medium.**

In order to simulate the effects of winds on the turbulence observations, we cut subimages from the 256x256 images and displaced them spatially by a constant number of pixels in the x and y dimensions. The analysis algorithms then were applied to compute the complex cross power spectral densities of the spectral images.

The basic algorithms used for the triangulation geometry as well as the successive frame analysis in the vertical direction is the cross power spectral density algorithm group first applied to geophysical phenomena by Gossard et. al.<sup>4</sup> The Cross Power Spectra for either a set

of one dimensional vector quantities  $V1$ ,  $V2$ , or a sequence of two dimensional images  $I1$ ,  $I2$  may be generically expressed as:

$$XP(V1_{i,}, V2_{i,} \text{ or } I1_{ij}, I2_{ij}) = \text{CFT}(V1 + iV2, \text{ or } I1 + iI2) \quad (2)$$

Where the data vectors or image fields have indices  $i$ , and  $i,j$  respectively

**CFT** is the complex Fourier Transform with the vector or image quantities combined as a complex argument.

Whereas Gossard et al described the process in terms of the cross correlation functions of a data series, modern algorithms such as the Fast Fourier Transform allow direct computation of the data set. This analytical procedure was implemented by using standard complex Fourier Transform Algorithms contained in Mathcad.<sup>5</sup>

### **3.1.2 Simulation of Triangulation Geometry**

We simulated the triangulation geometry in two dimensions corresponding to the right angled intersection of the low altitude structure and wind experiment described later.

### **3.1.3 Signal Analysis of Simulations**

We analyzed the simulation experiments for two cases. The 2-D single beam, vertical looking geometry that corresponds to the low altitude experiments measures MWIR scattered sunlight as a signal as the turbulent scattering irregularities are convected through the field of view of the sensors. The 2-D low altitude triangulation geometry corresponds to the scattered sunlight signal in the region of intersection of the sensor fields of view.

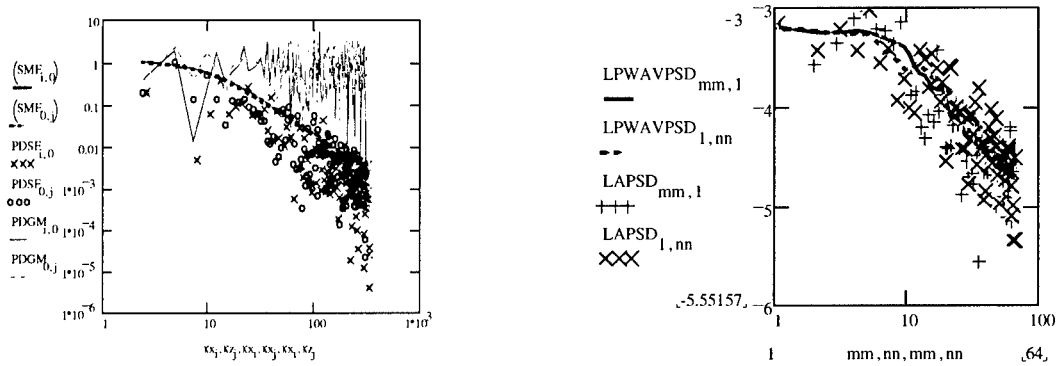
In the case of the vertical viewing sensor simulation, we computed the complex cross spectral information in both real and imaginary regime and using standard signal processing techniques (see reference 4) derived the AS (Amplitude spectra), PSD (power spectral densities), IPSD (imaginary components of PSD), and the PHSD (Phase Spectra) for each pair of images. Then we looked for peaks in the phase spectra (PHSD) which might correspond to the motion of the turbulent scattered scenes across the sensor field of view.

The two dimensional PSD's and Phase Spectra (PHSD) for an image array A are defined by the standard expressions involving the periodogram PA of the array A which is derived from the "cfft" complex Fourier Transform of the array A. In the simulation case, the image arrays A are the computed Gaussian matrices GM.

$$PA = \text{cfft}(A), \text{ PSD} = |PA|^2 \quad (3)$$

$$\text{PHSD} = -\arctan (\text{Im}(PA)/\text{Re}(PA)) \quad (4)$$

As a diagnostic we compare in Figure 3 the ideal Kolmogorov  $k^{-5/3}$  PSD input to the program, the computed periodogram for the convolution of the GM, Gaussian correlated noise matrix, and the raw periodogram of the array GM itself. The periodograms will be converted to power spectral densities (PSD's) later in the analysis

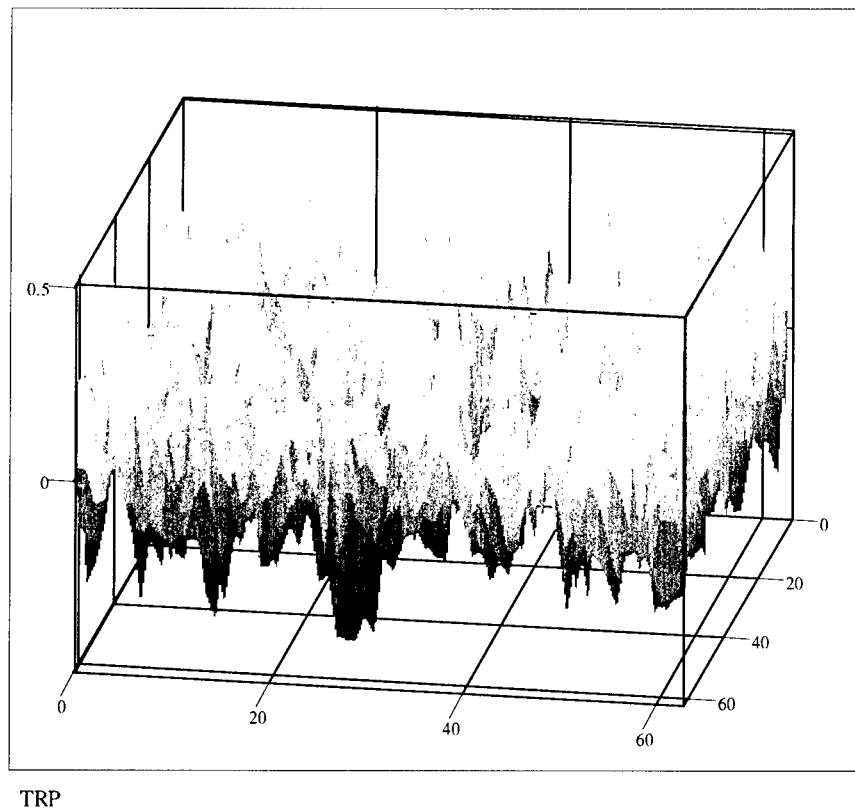


**Figure 3 Left Image Compares the Correlated Gaussian Noise Matrix 1-Dimensional Periodograms and the input data (smooth line). SME is the input PSD, PDSE is the Periodogram of the Noised Gaussian Matrix, and PDGM is the Periodogram of the Gaussian Noise Matrix before convolution with SME. Values of PDSE and SME are normalized to their maximum values. The right image shows the 1-D PSD's (lines) after a Parzen filter is applied. The data points correspond to the noisy periodogram in the left image. The x axes correspond to spatial frequency in units of inverse pixels.**

This exercise confirms that the PSD of the simulated random turbulence field reproduces the input spectrum.

Also it is necessary to validate that phase spectra can be produced, using the Gaussian random structure simulation. Figure 4 illustrates the phase spectrum over the range of zero to

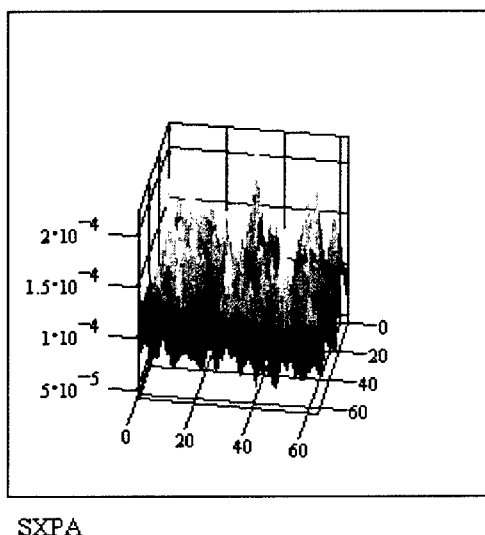
1/64 pixel spatial frequency range. Histograms indicate that the phase spectral field is a correlated Gaussian distribution as expected.



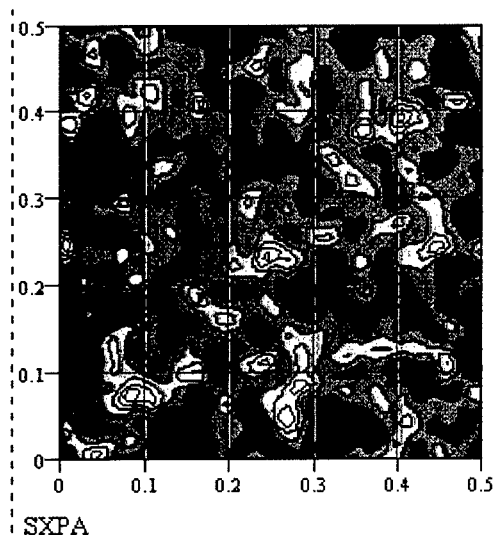
**Figure 4 Phase Spectrum for Gaussian Correlated Simulation Scene. The spectrum extends from zero to 1/64 pixel spatial frequency in both spatial dimensions and from -1 to 1 radian in phase.**

The Cross Power Spectral Density and Cross-Phase spectra are used to estimate the spatial displacement of the images and the vector velocity of the passage through the simulated two fields of view, from successive analyses of the sub-image pairs that are separated by 20 pixels.

The 2-D Cross PSD is smoothed by the Parzen window technique and then converted to a 2-D correlation function by means of a complex FFT. Figures 5A and 5B illustrate the 2-D PSD as a 2-D plot, and as a pair of 1-D cuts along the axes.



**Figure 5A Smoothed Cross PSD as a surface plot. Note the occurrence of several spectral peaks**

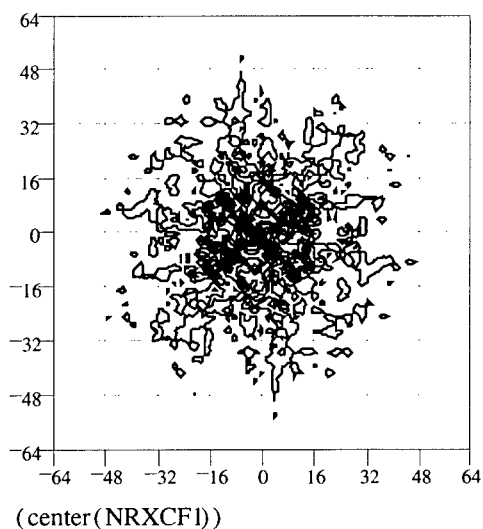


**Figure 5B Smoothed Cross PSD shows spectral peaks on the frequency x-y diagonal as anticipated. The other peaks appear to be extraneous features.**

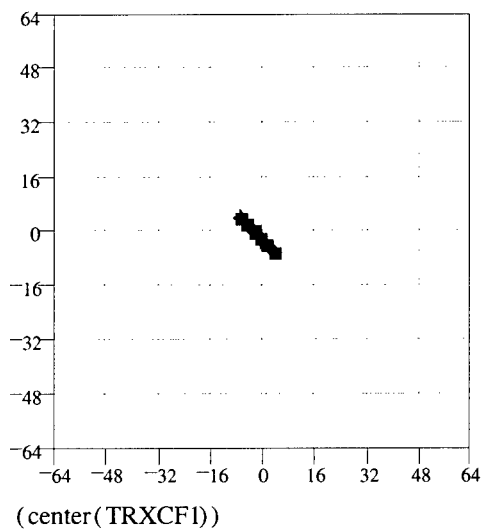
The coordinates of Figure 5A are in units of spatial frequency divided by the pixel dimension of the image (ie 128 pixels with 2 pixels being the Nyquist frequency). Units of 5B are in inverse pixels such that 0.5 is equivalent to the Nyquist frequency of  $1/(2 \text{ pixels})$ .

After conversion of the 2-D Cross PSD's to cross correlation plots, it is necessary to threshold the data to bring out the pertinent features on the desired scale. Figures 6A and 6B show respectively the un-thresholded 2-D cross correlation plot and the thresholded image. The threshold chosen for this case was 7 times the standard deviation of the cross correlation scene after removal of the random energy component at zero delay.

The axes are in units of  $1/\text{pixels}$ , hence the value 0.5 is equivalent to 2 pixel displacement and 0.1 is equivalent to 10 pixels displacement. The length of the thresholded correlation value in Figure 6B shows that the width of the correlation results equals about 20 pixels and is in the diagonal dimension as specified by the original image pairs.

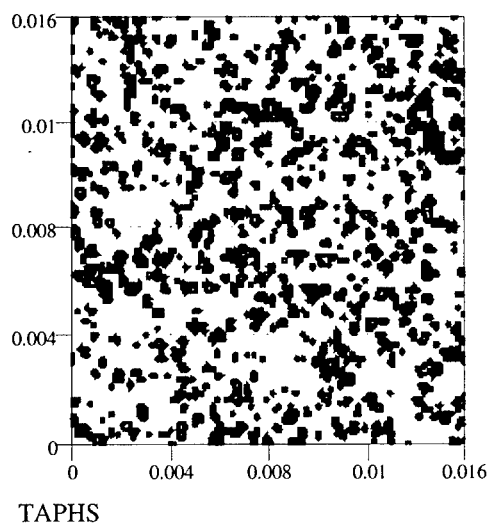


**Figure 6A 2D Cross Correlation Plot Unthresholded**

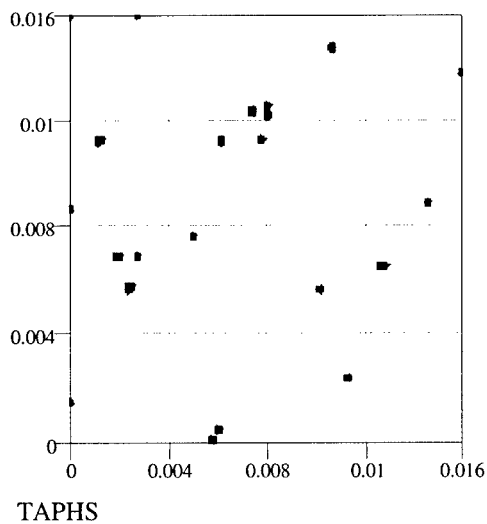


**Figure 6B 2-D Correlation Plot Thresholded at 7 Sigma**

Figures 7A and 7B show the phase spectrum thresholded at two levels, guided by the histogram of phase (not shown here). Spatial frequencies are in units of inverse pixels.



**Figure 7A Exceedance Map for 1 sigma Thresholded Phase Difference Spectrum.**



**Figure 7B Exceedance Map for 3 sigma Thresholded Phase Difference Spectrum.**

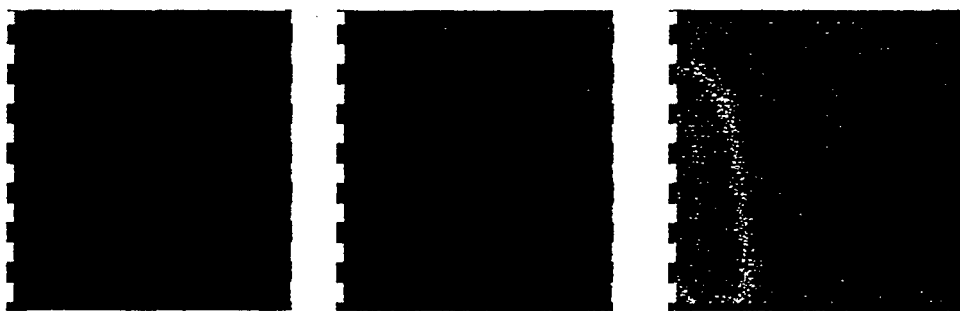


## VALIDATION BY EXPERIMENT

### 3.1.4 Low Altitude Wind and Turbulence Experiments

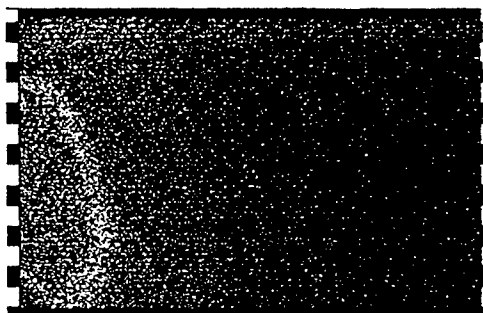
Low altitude wind and turbulence experiments were conducted at Kestrel Corporation using a set of filtered CCD cameras. These cameras are normally operated as an airborne remote sensing system by Kestrel Corporation and provide digital images with 8 bit resolution and a  $252 \times 252$  image pixel frame size. The goal of these experiments was to measure solar backscatter from aerosols and Rayleigh scatter in the lower troposphere, in the near vicinity (a few hundred meters) of the sensors. Appendix 1 summarizes the sky-scatter structure experiments conducted by Kestrel Corporation. In summary, we obtained individual green and blue images with unpolarized, horizontally, and vertically polarized images, plus dark response on each focal plane (cover on), and uniformity of the focal plane responsivity by use of a sunlit diffusion source.

Figures 8A to 8C illustrate luminance images (combined green and blue filters) used to determine the focal plane background dark noise (8A), the focal plane response to a diffuse illumination source (8B), and the response to the sunlit sky. In the case of the sunlit sky figure 8C illustrates the solar scatter symmetry very well in both green and blue. Unfortunately black and white reproduction is unable to show these features clearly.



**Figure 8A Dark Focal Plane**   **Figure 8B Diffusely Illuminated Image**   **Figure 8C Luminance image of sky showing green and blue scatter from the sun.**

Figures 9A and 9B show the solar scatter pattern observed by camera V2 observed in green and blue channels respectively. Image analysis can be accomplished using the composite G+B to enhance signal to noise ratio or the separated G and B channels.



**Figure 9A Solar Scatter Observed in Green Band, Sensor V2**



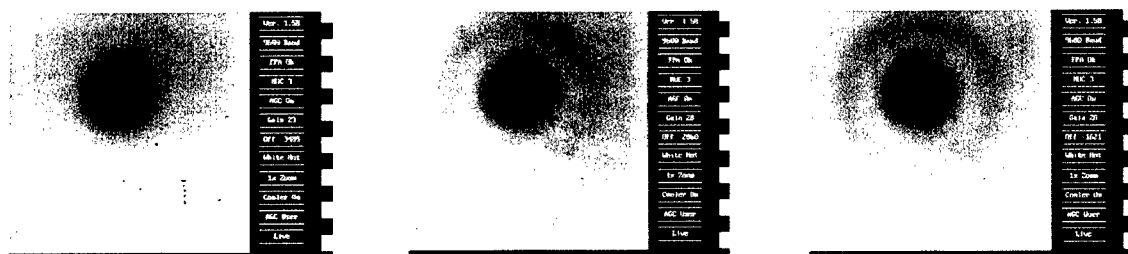
**Figure 9B Solar Scatter Observed Blue Band, Horizontally Polarized**

Several diagnostic configurations were explored for this experiment. The dark current statistical noise characteristics for the cameras were measured by covering the instruments and recording data. The focal plane structure noise was measured by putting an (almost) uniform diffusive surface over the lenses and recording data. Finally measurements were made in several wavebands, and in two polarization modes for vertically viewing sensors and for the virtual triangulation geometry with the sensors separated by 66.5 feet and the altitude of intersection about 100 feet. Table 1 summarizes the focal plane and objective-space characteristics of the visible and NIR cameras. For more details of this experiment refer to the Kestrel data summary report in Appendix 1.

**Table 1**  
**Camera Characteristics and Projected Objective Space Quantities**

Camera	Band	Pixel IFOV mrad	FOV mrad	Pixel Footprint At 100 ft	Image Dimension – pixels
V2	Green,Blue (Polarized)	.91	442 x 1075	~ 2.8 cm	768x486
V4	Green,Blue (Polarized)	.91	442 x 1075	~ 2.8 cm	768x486
IR	MWIR 3-5 mm	1.4	680 x 680	~ 4.7 cm	486 x 486

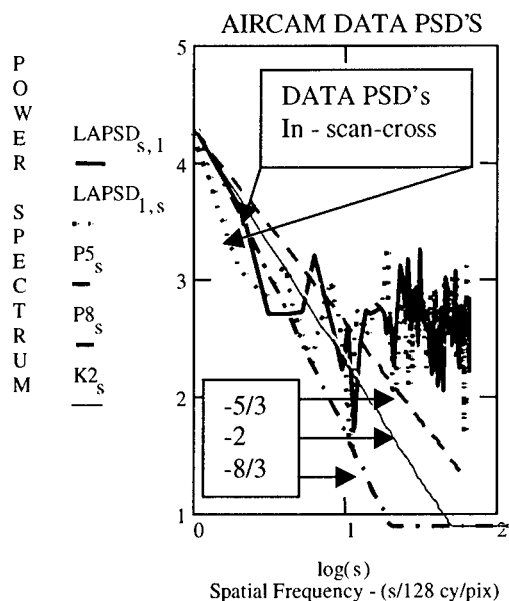
In addition to the visible G-B data sets an Amber near infrared (3-5  $\mu\text{m}$ ) camera was operated to view zenith in the vicinity of the visible beam intersection. This camera was used primarily to see if the signal to noise ratio on a cooled MWIR system could detect structure in the lower atmosphere. Because of the 8-bit digitization level (i.e. 256 intensity levels) we believed that this sensor might be marginal for detecting atmospheric structure but the presence of water vapor and nearly subvisual clouds may have enhanced our ability to detect turbulence and winds. Figures 10A – 10C compares one data frame obtained by the IR camera with the focal plane response to both cold and hot plates placed over the objective lens. It is apparent that a strong asymmetry in focal plane response exists.



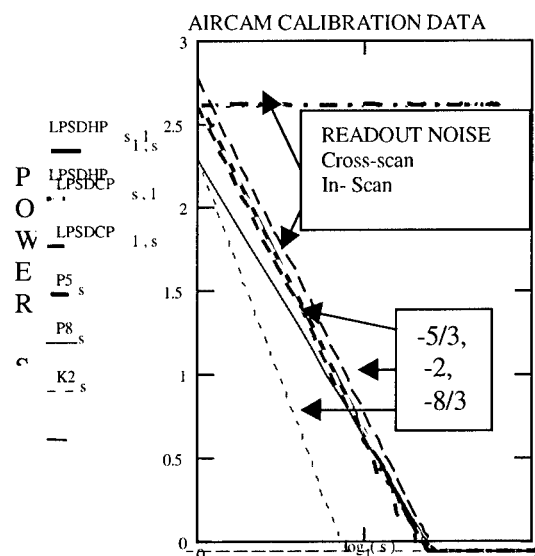
**Figure 10A IR Data Image    Figure 10B IR Hot Plate Image    Figure 10C IR Cold Plate**

The 2-dimensional spatial PSD's (Power Spectral Density estimates) of truncated images (128x128) pixels and 2-d correlation functions were computed. The images were truncated to ensure that radial variation in focal plane noise and responsivity could be minimized. This procedure also (nearly) eliminates the requirement for flat-fielding the data frames.

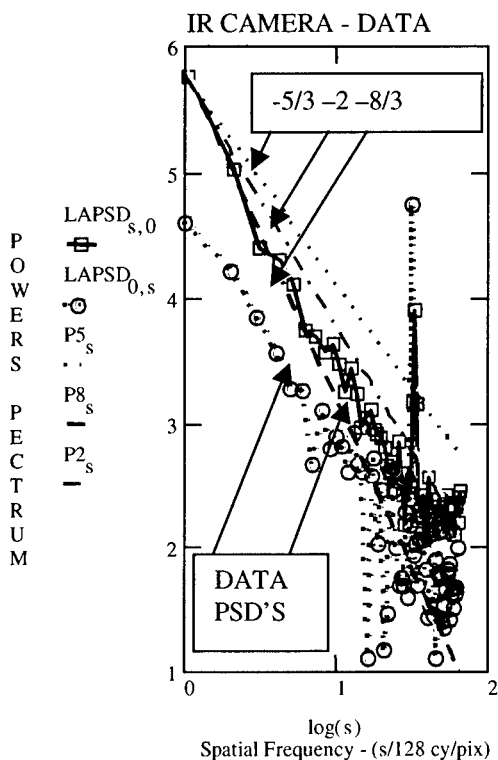
The PSD's computed for the Kestrel Aircam imager in measurement mode, dark current mode, and diffuse scatter mode are compared in Figure 11A and 11B. Figure 11A compares the measured sky background spectrum with example spectral slopes,  $k^{-5/3}$ ,  $k^{-2}$  and  $k^{-8/3}$ . The experiment spectrum drops into the noise level at a spatial wavelength about 30 pixels. Figure 11B shows that the focal plane pattern and dark current noise (symmetric in  $k$ -space) and the readout noise ( $1/f$  or  $(k^{-2})$  spectrum in the readout direction may be a significant deterrent to measurement of very small-scale structure inasmuch as the structure spatial frequency may lie beyond the noise cutoff frequency.



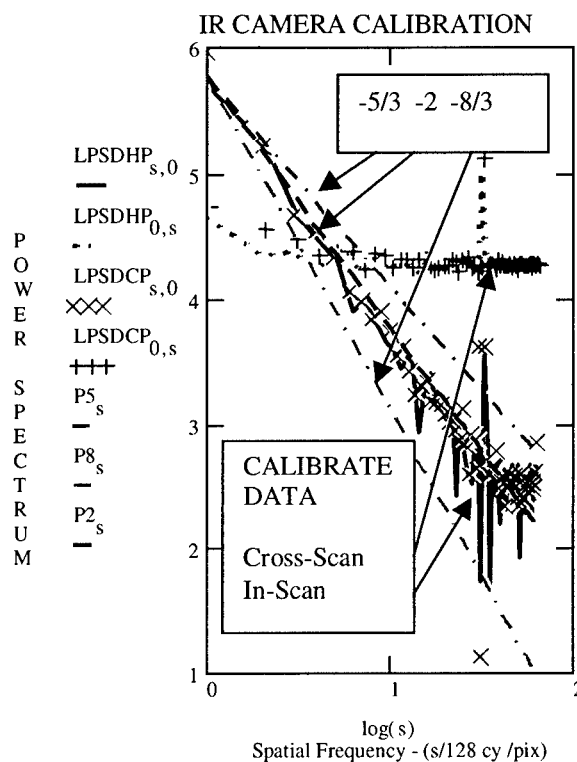
**Figure 11A AIRCAM sky data 1-D PSD compared to slopes,  $-5/3$ ,  $-2$ ,  $-8/3$ . Note that this camera was uncooled.**



**Figure 11B AIRCAM Calibration PSD's for dark counts and uniform illumination The readout noise is the k-2 slope**



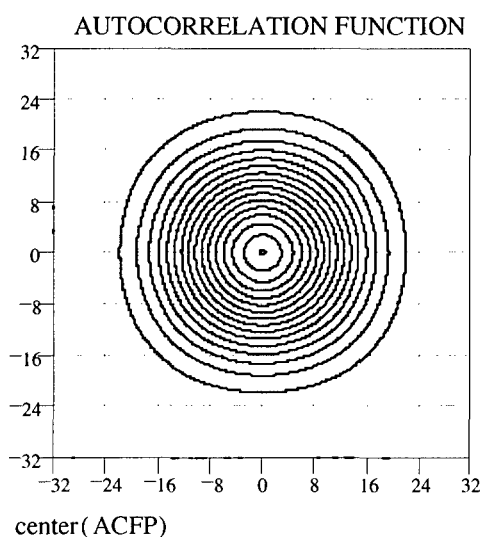
**Figure 12A IR Camera Data PSD's Compared to model PSD slopes.**



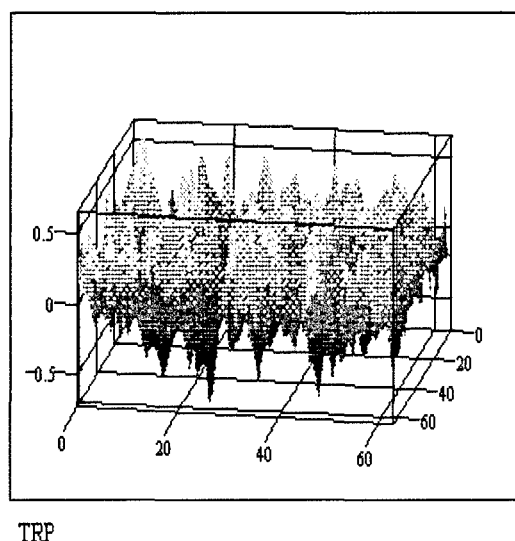
**Figure 12B IR Camera PSD's on Calibration Plates compared to models**

Figures 11 and 12 clearly show the role of focal plane noise and readout noise in the AIRCAM and IR sensors. The readout noise is clearly  $1/f$  (in amplitude) and is approximately the same magnitude for the IR sensor sky data as for the hot plate and cold plate calibration frames. In fact, the mean value of sky background in the IR band was lower than either the hot or cold calibration frame values, indicating that the sky was colder than either calibration temperature. For these reasons we do not attempt to correct the IR camera data for noise and cannot pursue analysis in terms of structure motion and correlation length.

We pursued estimation of the phase drift and the correlation function parameters for the AIRCAM data based upon the nearly 2 order of magnitude signal to noise ratio ( accounting for focal plane plus readout noise spectra). Figure 13 illustrates the average autocorrelation function for the five frames of data. Figure 14 presents the “auto-Phase spectrum for the same set of AIRCAM data frames.



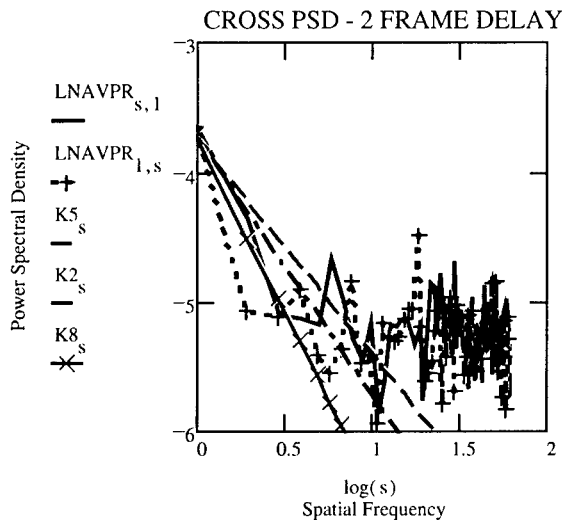
**Figure 13 Autocorrelation Function for 5 successive AIRCAM Image Frames**



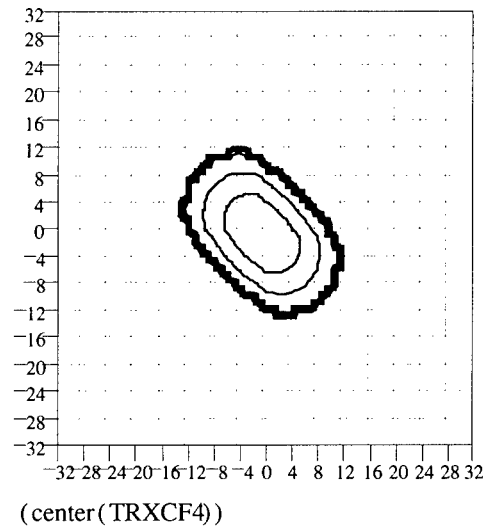
**Figure 14 Average Phase Spectrum for Same set of AIRCAM Images**

The autocorrelation data clearly shows a symmetric pattern characteristic of the Gaussian (presumably) characteristic of atmospheric low altitude turbulence as averaged over 5 successive image frames. The phase data are ambiguous, but seem to indicate definite phase peaks at selected spatial frequencies which may be emphasized by our thresholding method.

The Parzen- windowed power spectra illustrated in Figure 15A are used to compute the cross correlation functions for successive image frame delay times as illustrated in Figure 15B.



**Figure 15A Cross PSD for Successive AIRCAM IMAGE Frames. Straight Lines are K-5/3, k-2, and k-8/3 slopes.**



**Figure 15B Cross Correlation Function For Same Image Frames. Displacement From center indicates a 2 pixel lag in Diagonal direction (=2 mrad).**

The results shown in Figure 15A and B show that valid cross PSD and correlation functions can be computed for atmospheric turbulence on a small scale despite the high noise levels in the sensor and backgrounds. The 2 pixel offset in the correlation function center corresponds to a horizontal “wind” displacement of the turbulent structure of about 40 cm corresponding to a wind velocity of about 5 to 10 mph. This is consistent with local observations.

### 3.1.5 High Altitude Experiments

Two types of high altitude simulation experiments were conducted. We analyzed the GLO spectrometric data from the University of Arizona instruments flown on Shuttle Flight STS-85. The GLO instruments were flown in the look-forward, look-back configuration with a  $\pm 65$  degree azimuth with respect to the perpendicular to the spacecraft velocity vector to provide virtual triangulation geometry. The instrument lines of sight intersected the limb at about 58 km.

The second experiment was observation of the airglow emissions in the visible to Near-IR spectral range using the Kestrel Fourier Transform Hyperspectral Imager (FTHSI). In both experiments, the atmospheric airglow emission was the target signal. The goals of these experiments were first, to determine if the virtual triangulation geometry actually could be implemented by space-borne detectors and second, to determine if an FTHSI instrument could produce useable atmospheric data with sufficient spatial and spectral resolution to identify well known atmospheric airglow features.

### 3.1.5.1 GLO Airglow Data Analysis

Dr. Lyle Broadfoot of University of Arizona, Lunar and Planetary Institute has conducted GLO experiments on several STS flights. A selection of data from STS - 85 were provided to this project under subcontract. The geometry of the STS - 85 experiment is similar to that shown in figure 1 except that the two sensors, GLO 5 and GLO 6 are oriented to intersect the limb with a line-of-sight tangent altitude of 59 km. Appendix 2 contains selected details of the STS - 85 GLO experiments. Table 2 summarizes the spacecraft parameters and camera details relevant to this experiment

**Table 2**  
**Spacecraft Parameters and Camera Characteristics**

<b><u>Spacecraft:</u></b>	<b>STS - 85</b>
Altitude:	400 km
Orientation:	Pitch = 180 degrees, Yaw = 270
Velocity	~ 7 km/sec
<b><u>Hyperspectral Cameras:</u></b>	<b>GLO 5 &amp; GLO 6</b>
Camera Spectral Range:	
Camera Orientation:	Elevation Angle (depression) 22 degrees, azimuth = $\pm 65$
degrees	
LOS Tangent Altitude (center)	59 km (drifts with time)
Range to Tangent:	2120 km
LOS Velocity at tangent	~ 5 km/sec
Geometrical Time Lag of 5-6 LOS	~420 sec. (= 2120 km/5 km/sec )
Spatial footprint	5 km at earthlimb

Figure 16 summarizes the GLO airglow data for one satellite pass. Figures 17- 21 show the detailed GLO measurements of four airglow species emissions where the time variability between GLO 5 (forward-looking) and GLO 6 (backward-looking) and the time delay

between the data sets is established by matching intensity features. Each plot shows a delay from the forward to backward-pointing sensors appropriate to the airglow enhancements at their emission altitude. The thin airglow emitting layers are intercepted by the sensors operating at 59 km tangent altitude at two positions, a near field between the tangent point and the sensors and a far field intercept point on the far side of the 59 km tangent point. The emission “onion skin” characteristic leads to two possible lag times, corresponding to features in the near and far fields. It is unlikely that the sensor vertical resolution is sufficient to spatially resolve the far-field component so we will concentrate on near field results.

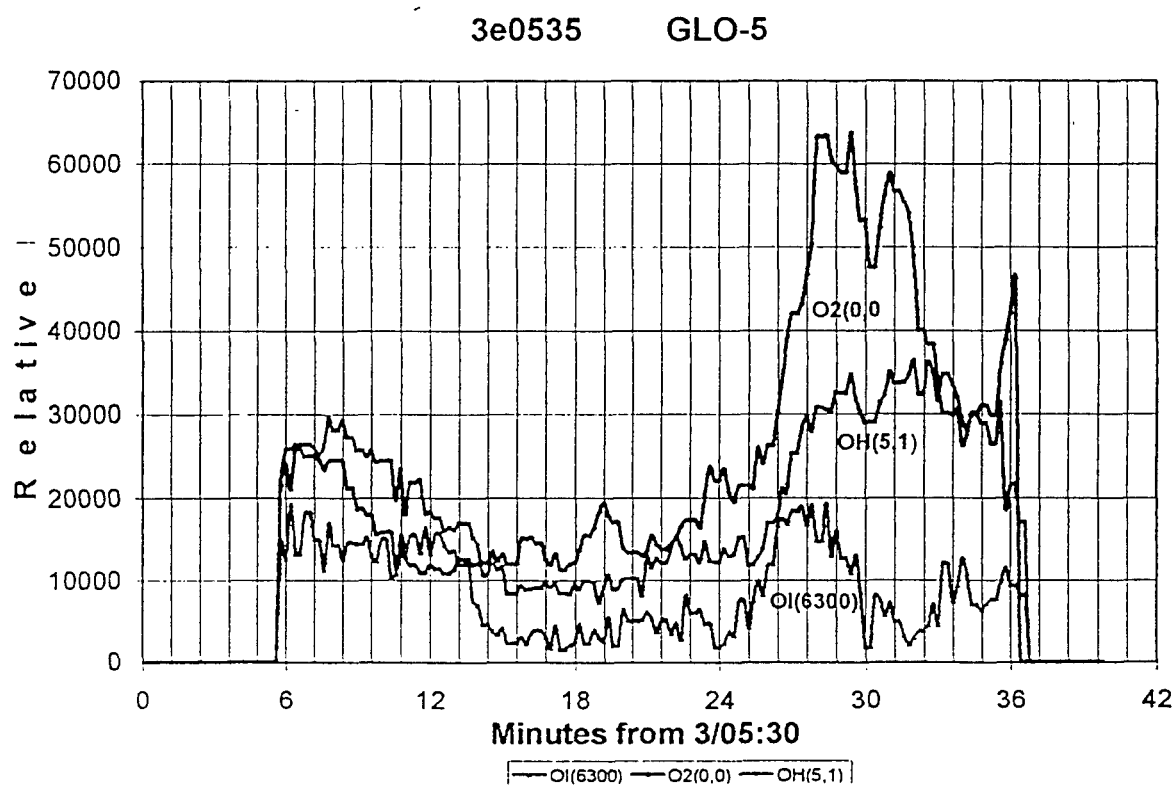
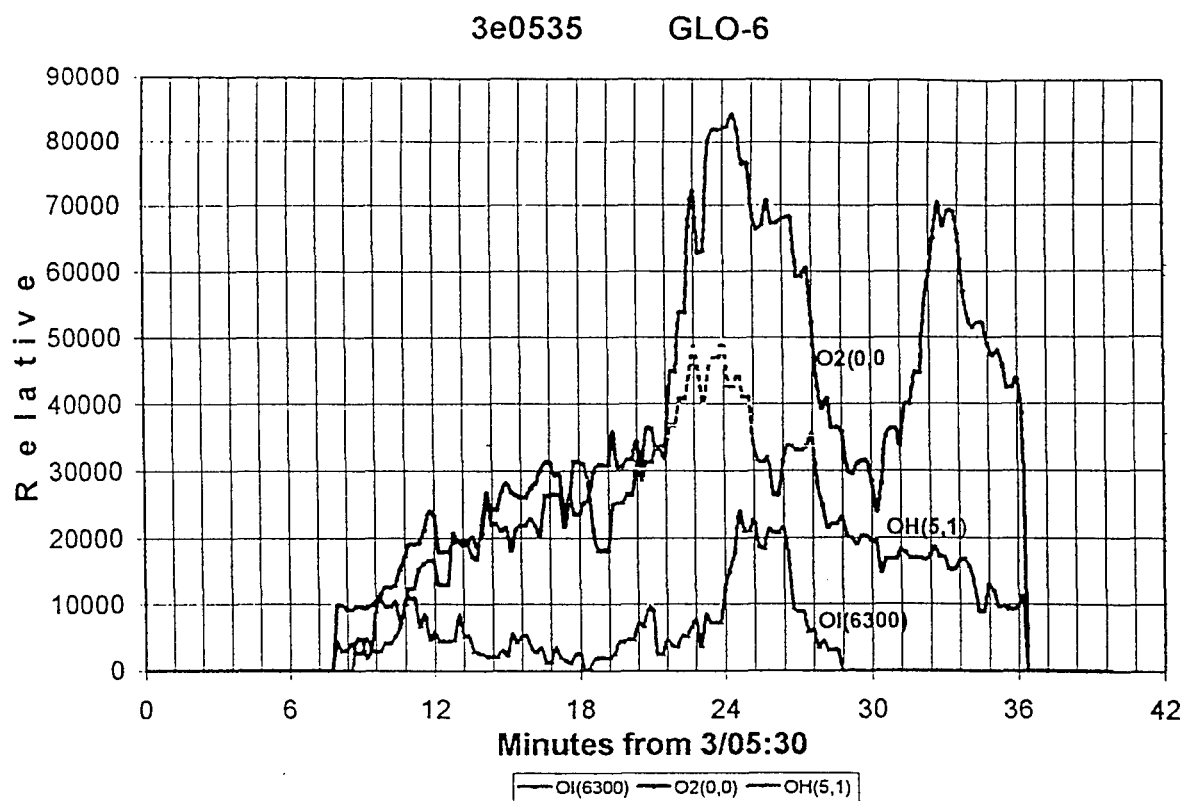
We computed the delay times for a number of emission altitudes: 85, 95, 130 and 240 km to compare with the GLO5-6 delay observations. Table 3 summarizes the airglow characteristics and near- and far-field ranges for the sensor to intercept the emitting layer. The corresponding delay times compared with the GLO values. We adopted an LOS velocity of 6 km/sec.

**Table 3**  
**Airglow Delay Predictions and Observations**

<b>Emitter Species</b>	<b>Band nm</b>	<b>Altitude km</b>	<b>Range km –Near</b>	<b>Range km – Far</b>	<b>Delay min. –near</b>	<b>Delay min.-far</b>	<b>GLO Observation- min.</b>
OH	762	85	670	6100	1.9	17	7.2
Na	589.3	95	470	6300	1.3	18	4.8
O2	762	95	470	6300	1.3	18	2.8
OI	557.7	~130	340	6400	1.	18	2.8
OI	630	~240	320	7000	0.9	19	2

Comparison of the computed estimates of delay time vs. GLO measurements shows that most of the observed delay times are about a factor of 2 greater than the predicted estimates. Some of the factors creating this discrepancy may be due to airglow horizontal structure, and the exact altitude of the layer as the two sensors scan the same region. There also is a trend in the GLO observations to much higher values of delay time in the OH and Na emissions. Both of these emissions are highly structured spatially and temporally so may not indeed be statistically stationary over the period of the measurement, about 30 minutes.





**Figure 16 Summary of Airglow Intensity vs. Time for GLO-5 and GLO-6**

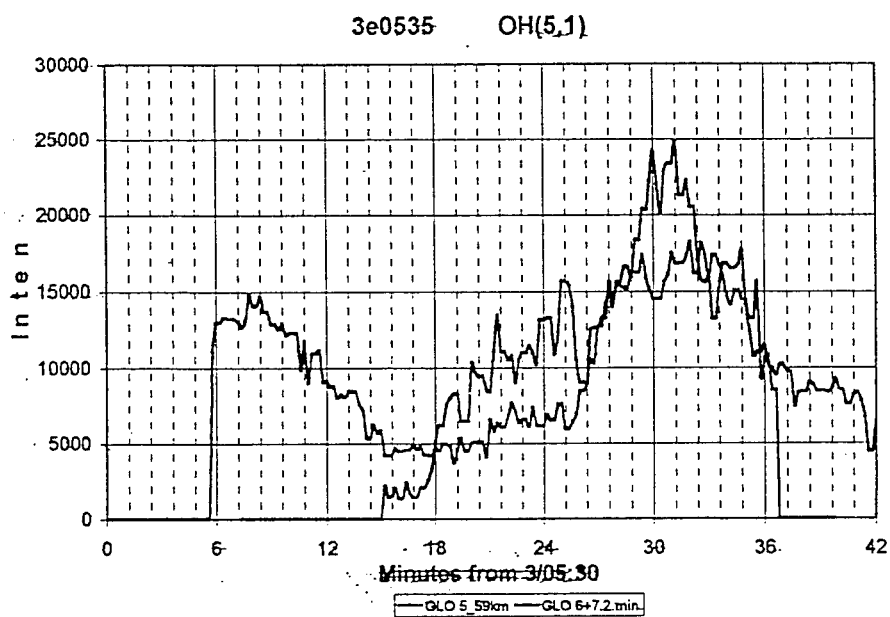
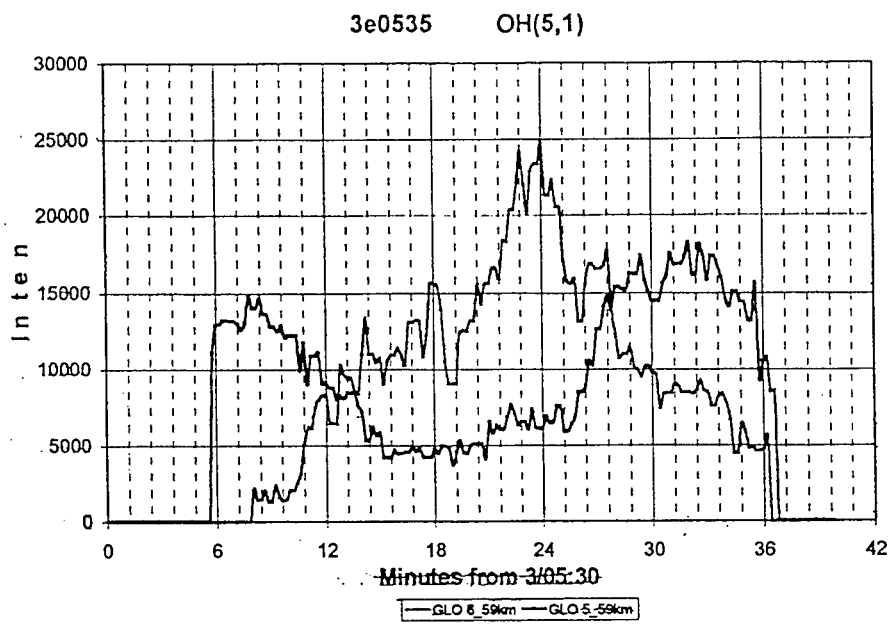
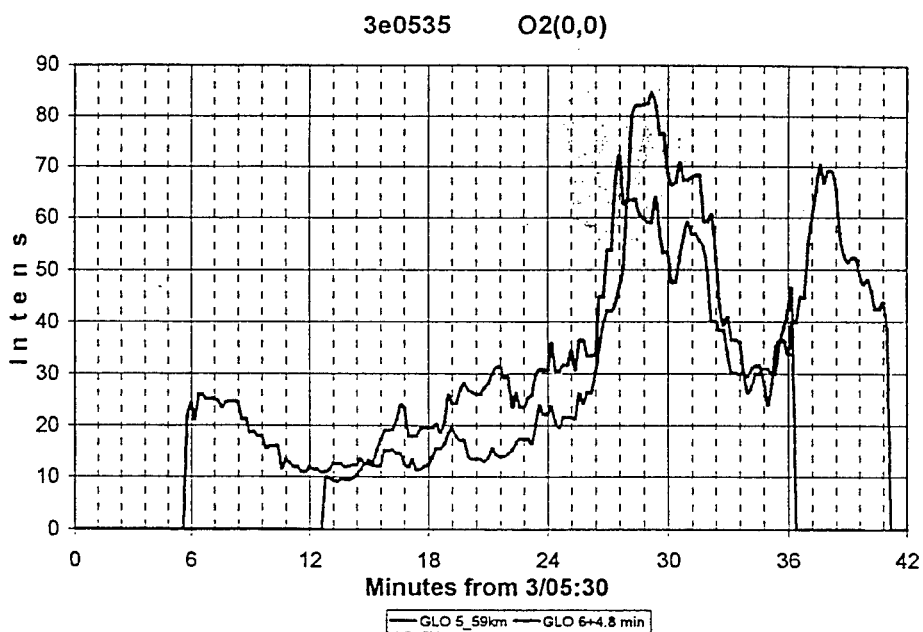
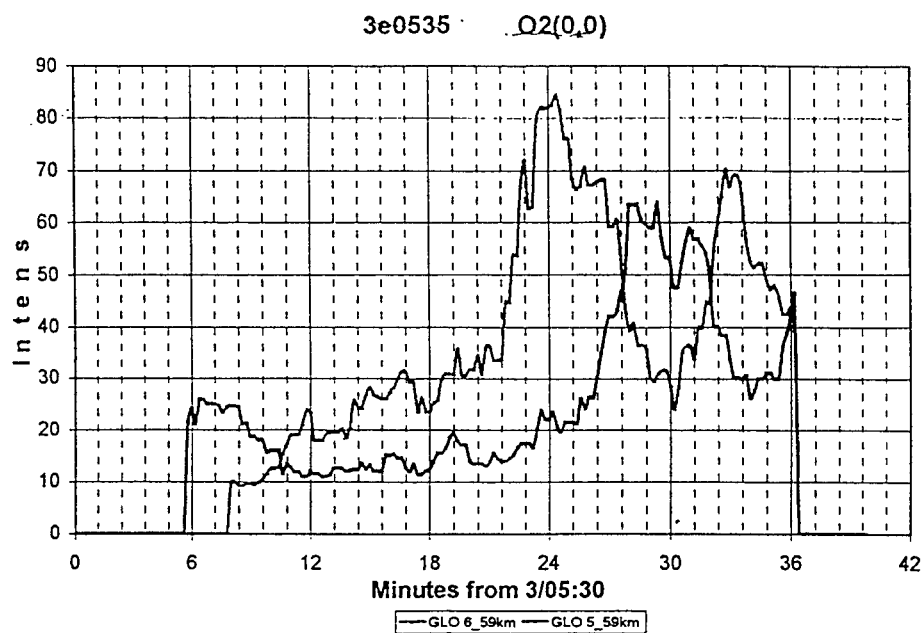


Figure 17 OH (5-1) Meinel Band Intensity vs. time from GLO-5 and GLO-6. The top panel shows the data at the time of acquisition. The bottom panel shows the same data with the trailing sensor plot time offset for a best fit of the intensity features



**Figure 18 O2 Atmospheric Band Intensity vs. time from GLO-5 and GLO-6. The top panel shows the data at the time of acquisition. The bottom panel shows the same data with the trailing sensor plot time offset for a best fit of the intensity features**

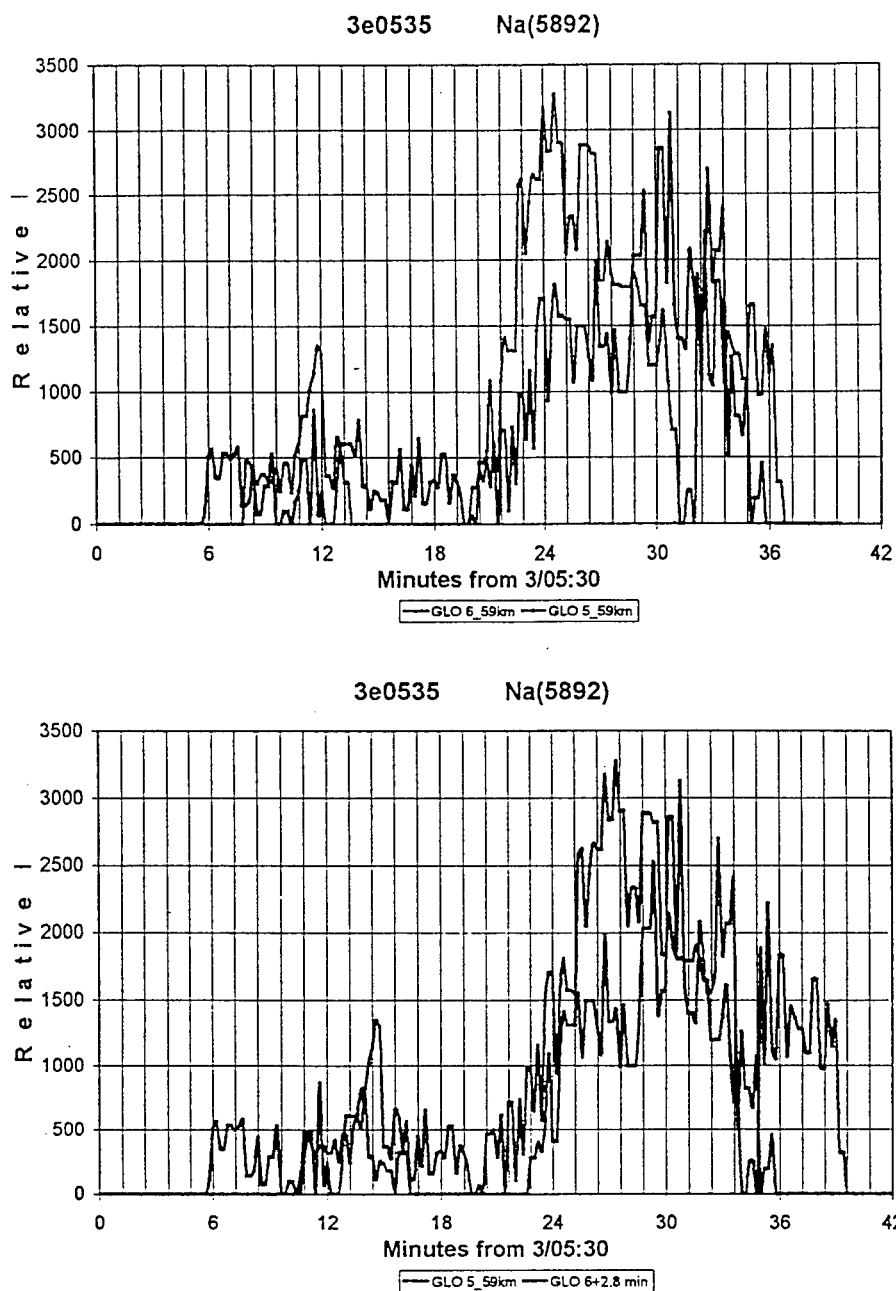
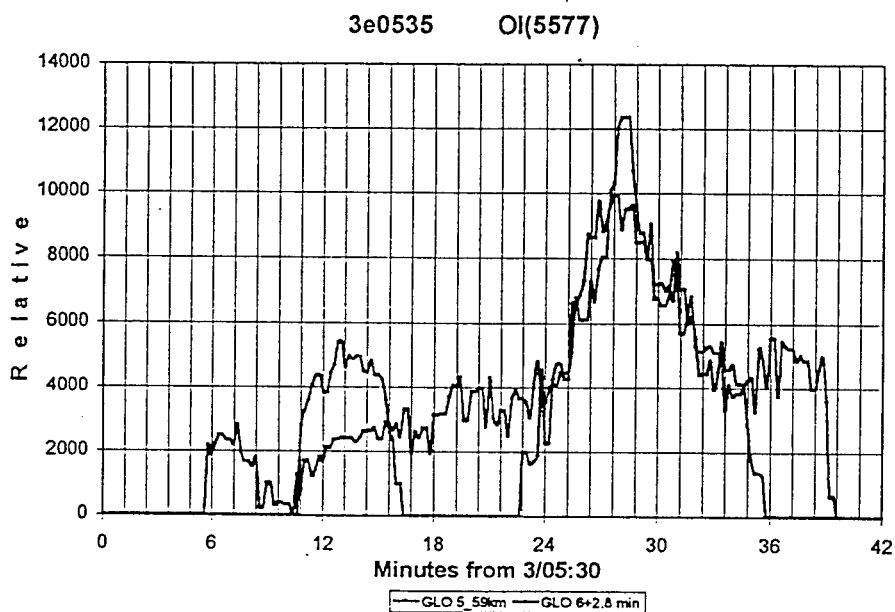
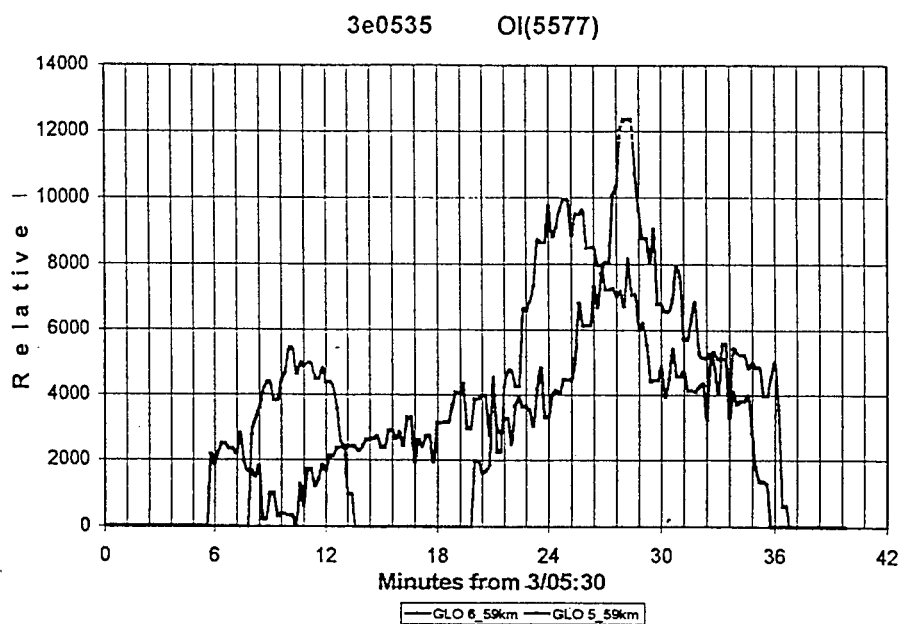


Figure 19 Na 589 & 589.6 nm Intensity vs. time from GLO-5 and GLO-6. The top panel shows the data at the time of acquisition. The bottom panel shows the same data with the trailing sensor plot time offset for a best fit of the intensity features



**Figure 20 OI (557.7 nm) line Intensity vs. time from GLO-5 and GLO-6. The top panel shows the data at the time of acquisition. The bottom panel shows the same data with the trailing sensor plot time offset for a best fit of the intensity features**

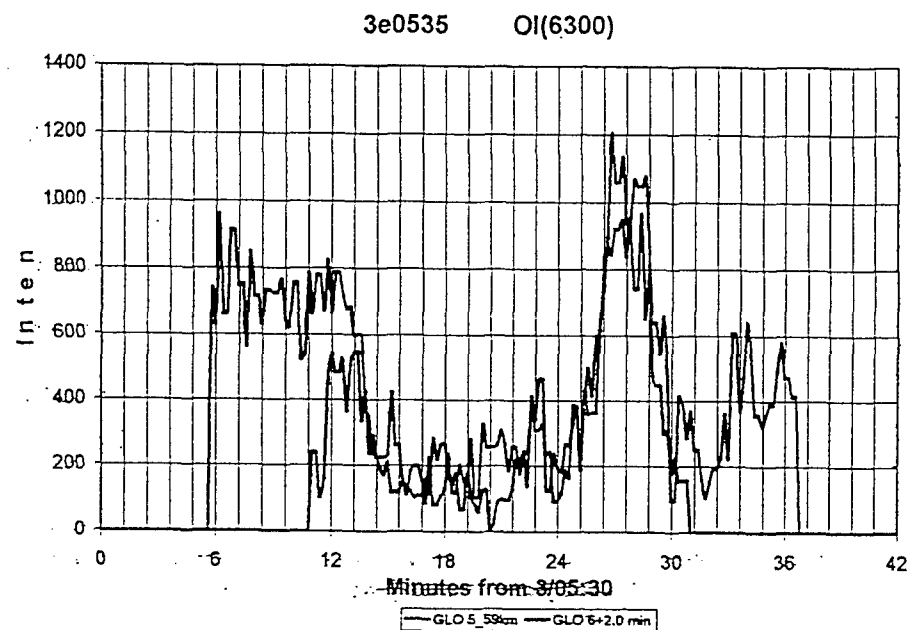
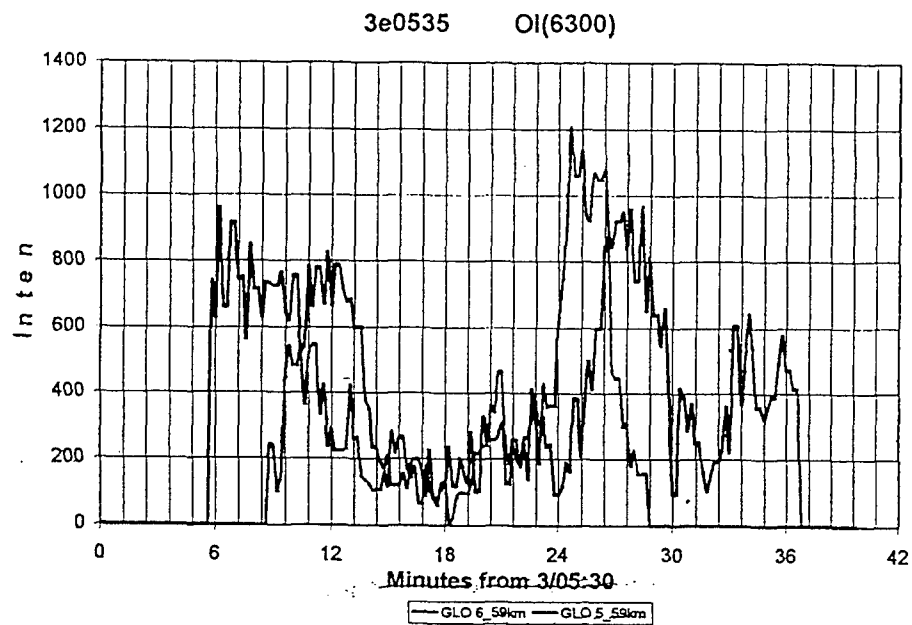


Figure 21 OI (630 nm) line Intensity vs. time from GLO-5 and GLO-6. The top panel shows the data at the time of acquisition. The bottom panel shows the same data with the trailing sensor plot time offset for a best fit of the intensity features

### 3.1.5.2 Hyperspectral Imager Analysis of Airglow Spectra

The Kestrel Fourier Transform Hyperspectral Imager (FTHSI) was used to estimate the capabilities of this instrument to measure high altitude airglow spectra in the 400 to 900 nm range from the ground. This instrument was not designed to perform this measurement mission, but we decided to attempt the measurement and to test the focal plane cooling requirement required to detect airglow.

Figure 22 illustrates the night airglow data obtained by the FTHSI operating in an out-of-design dry-ice cooling mode. The cross track data were integrated to obtain a higher signal to noise ratio to supplement the cooling effect. We found that the streetlight and other urban signals such as Na and Hg lines were observed, but also we could identify most of the emission features as airglow features. Figure 22 also identifies the airglow features including Na and Hg emissions from streetlights. Only a few radiance enhancements were not identified, but this is common in an urban background environment. The integration time for these measurements was order 1 second or less, (down to TV 30 Hz sample rate).

Figure 23 shows night airglow features measured many years ago as published in Chamberlin<sup>6</sup>

We conclude that the FTHSI and similar imaging spectrographic instruments have the sensitivity to detect airglow emissions in the visible-near IR bands and that useful space-borne measurements can be obtained provided that the focal planes are cooled sufficiently.





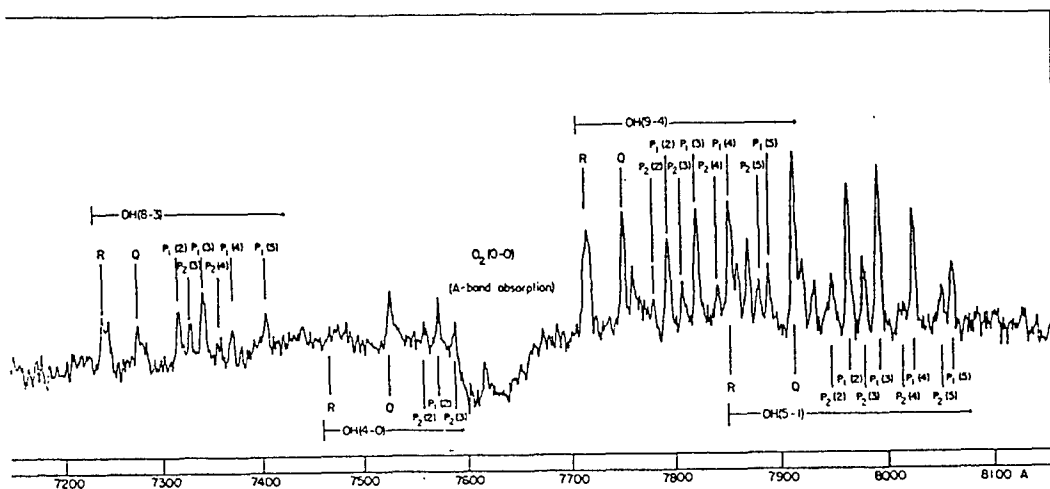


FIG. 9.11. Microphotometer tracing of nightglow spectrum, 7100-9000 Å. Plate dispersion, 70 Å/mm.

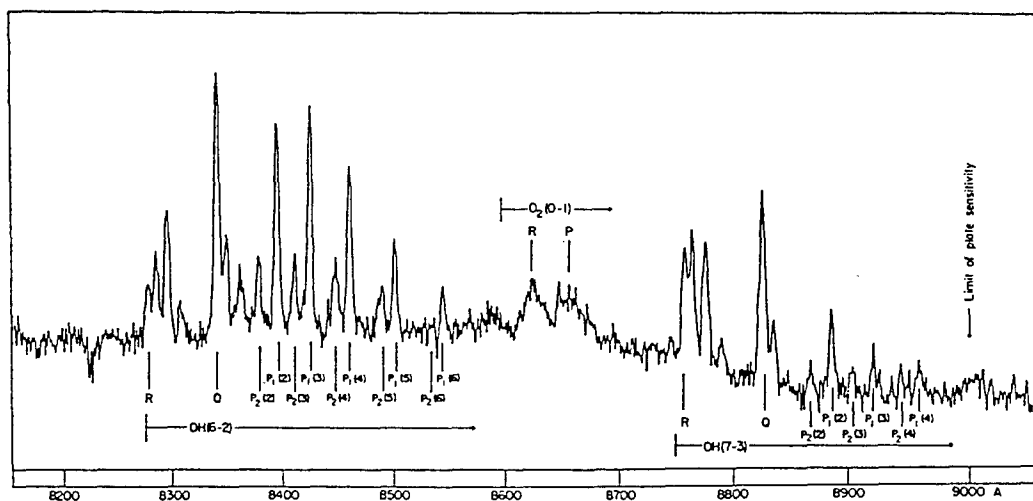


FIG. 9.11 (cont.)

After Chamberlain and Roesler [1955a]; courtesy University of Chicago Press.

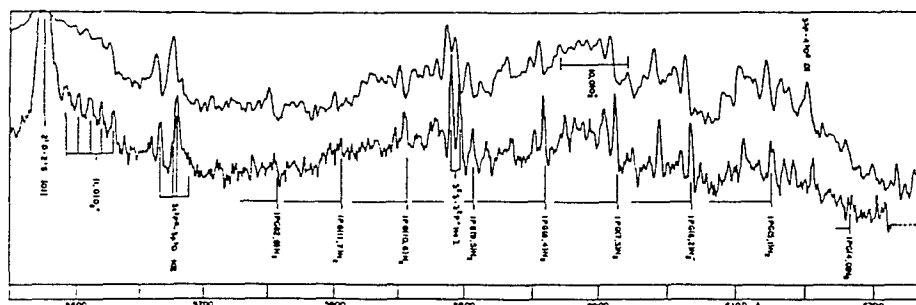


FIG. 5.16. Microphotometer tracings, 5550-6240 Å, of spectrum in Fig. 5.4. After Chamberlain and Meinel [1954a]; courtesy University of Chicago Press.

**Figure 23 Airglow Spectra published by Chamberlain for the visible, near-IR range.**

#### 4. SENSOR AND SPACECRAFT CONCEPT DEVELOPMENT

In order to provide an initial feasibility check regarding incorporating this instrument concept into a small satellite bus, we initiated a simple satellite design concept based upon the dimensions of the MightySat II bus. The purpose of this design effort was to ascertain whether off the shelf CCD cameras and hyperspectral imagers could be coupled with a MightySat II class buss and preserve the telescope resolution, instrument placement, and other parameters required for the virtual triangulation, hyperspectral measurement requirements. Figures 24 and 25 illustrate the design concept drawings.

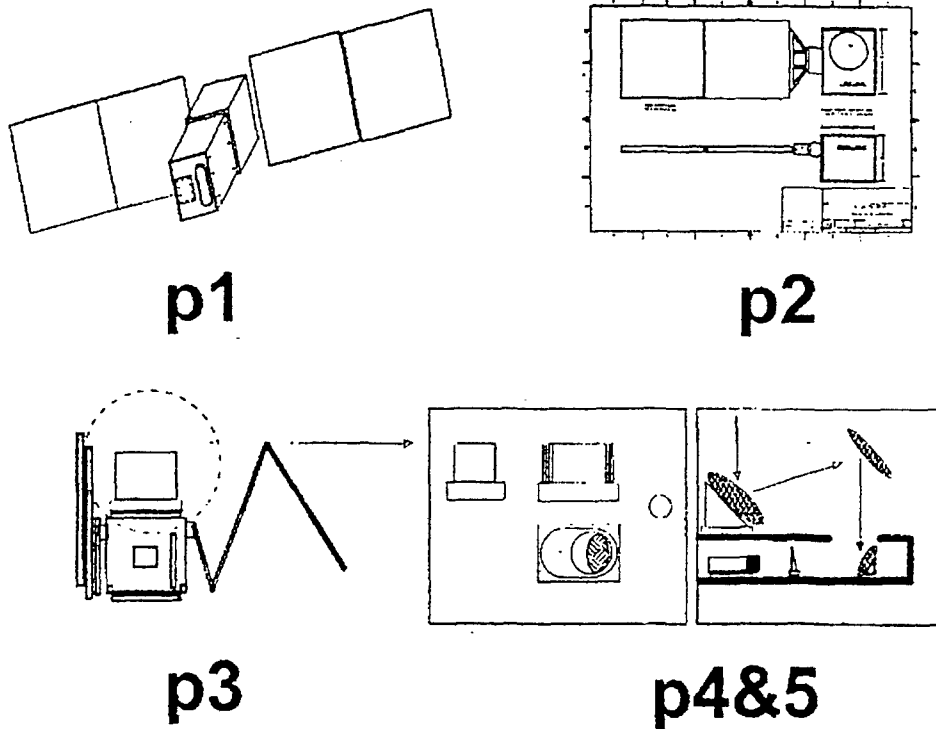
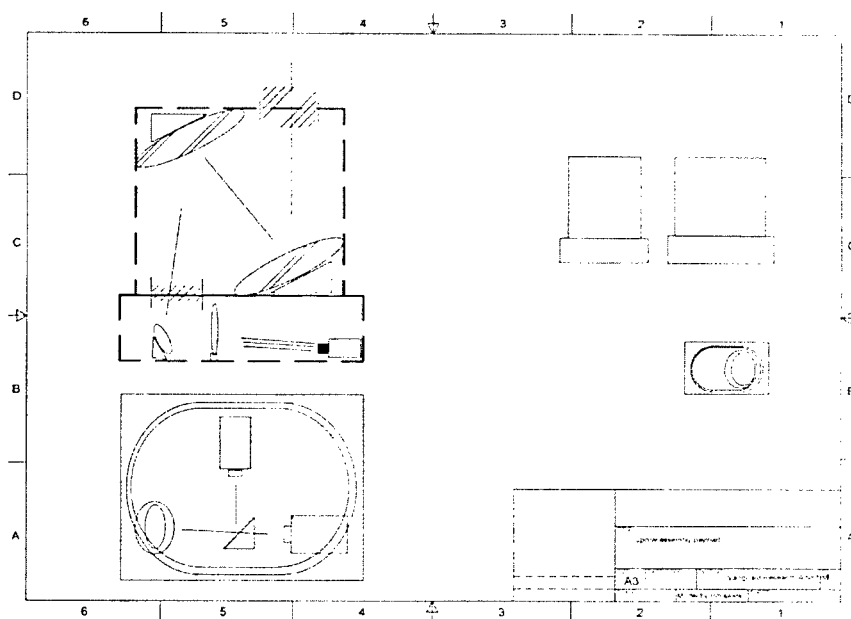


Figure 24 Overall line drawing of conceptual sensor and satellite bus based upon MightySat-II bus dimensions. The instrument dimensions are semi-scale, based upon commonly available cooled CCD focal plane detectors for UV and visible spectral range. An MWIR instrument may be somewhat larger and will require a second beam splitter.

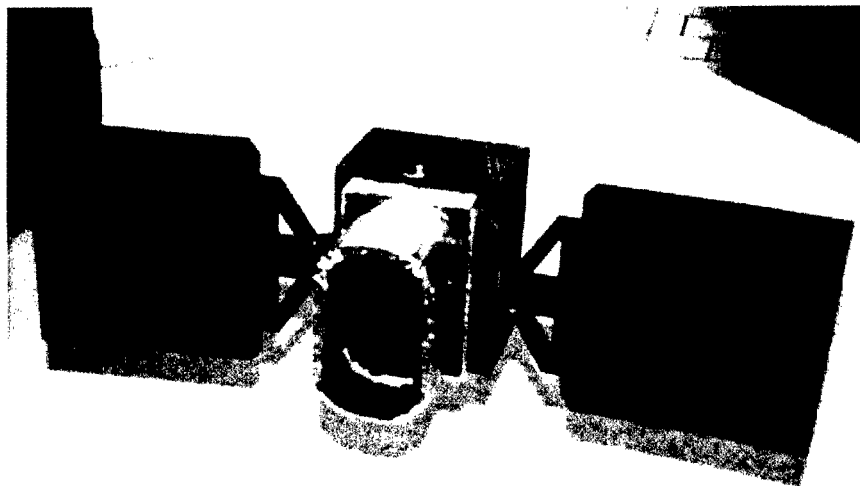
Panel P1 shows an overall view of the MightySat II bus, solar panels, and attached optical system. P2 shows orthogonal views of the concept satellite MightySat II bus. P3 shows the third orthogonal view with the solar panels partly folded. The optical system is designed to fit within the envelope of the folded solar panels, with the exception of the heat exchanger on the space-viewing side of the satellite. P4 shows the overall envelope of the proposed optical system the configuration for which is shown in P5. We plan to use dielectric layered beamsplitters to separate UV, visible and IR optical signals.

Figure 25 shows the optical layout of a two camera (UV and IR) sensor. The dimensions of the components are approximately to the scale of commercially available visible and UV cameras. A cooled MWIR camera such as those made by AMBER would be considerably larger, but the height of the optical module would fit would be scaled to accommodate the specific unit selected.

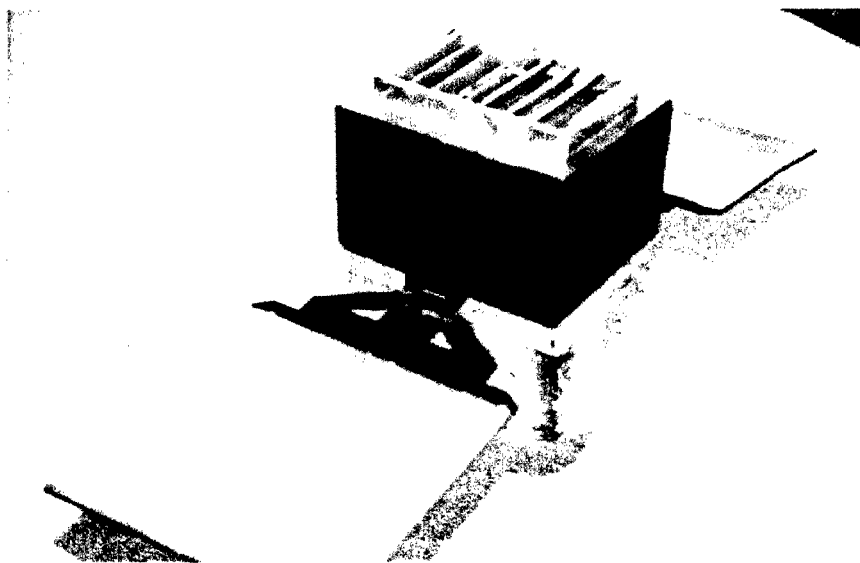


**Figure 25 Telescope mirror layout for conceptual instrument design. The instrument uses an off-axis telescope in order to maximize cross track field of view.**

We also assembled a very semi-scale model of a satellite to show that commonly available cooled CCD detectors would fit on a MightySat-II bus. Figures 26A and 26B show a rendition of the satellite concept as a third-scale model.



**Figure 26A Front view of concept satellite model based upon MightySat II dimensions. The optical sensor unit is a small module between the telescope and the satellite bus.**



**Figure 26B Rear view of concept satellite model as above. The focal plane and telescope-cooling grid is attached to the space-directed black surface of the MightySat-II bus. The present concept involves using heat pipes to cool the telescope and focal planes.**

## 5. COMMERCIAL APPLICATIONS

Commercial and military applications for the multispectral, virtual triangulation remote sensing technique lie in the following areas:

- Airborne sensing of atmospheric clear air turbulence (CAT) and small scale turbulence as applied to airborne and space-borne laser weapons systems,
- Satellite-borne mapping of atmospheric turbulence, waves, and winds as applied to transmission of energy by means of waves and turbulence from the troposphere through the stratosphere into the mesosphere,
- Satellite-borne sensing of atmospheric turbulence regions which would affect airborne and space-borne laser weapons systems.

Our commercialization plan incorporates the following elements which will be implemented in the Phase II program:

- Modification of the existing space-qualified GLO sensor system developed by University of Arizona, by improving the field of view, pixel resolution, and inclusion of a Fourier Transform Hyperspectral imager in place of the dispersive imaging spectrometer. We also will incorporate the quad C-40 processor. Both the FTHSI and the Quad C-40 will be tested on the MightySat II.1 satellite testbed. This instrument will operate in the UV through NIR spectral range (300 to 900 nm). This instrument will be space qualified for Shuttle flight. We will seek a Shuttle manifest slot in collaboration with our subcontractors at University of Arizona.
- Development of a GLO and FTHSI-based instrument which will be compatible with the MightySat II spacecraft platform. We will seek a MightySat II satellite development and launch slot in collaboration with the USAF Research Laboratory Spacecraft Division.
- Exercise of the developmental prototype instruments on an airborne platform to investigate their capabilities for detection of CAT.

## 6. CONCLUSIONS

We summarize the work performed and the accomplishments on this SBIR contract as follows:

- Partial simulations of the background Gaussian correlated random noise structure allowed measurement of the PSD and Phase Spectra and velocity for the virtual triangulation geometry concept. High signal to noise ratio in these simulations and subsequent measurements is required to accomplish very high spatial resolution atmospheric structure measurements from satellite altitudes.
- More detailed simulations, involving a true 3-D correlated random background drifting through the triangulation sensors' fields of view are required and could not be performed in this phase because of computer memory limitations.
- Ground-based CCD camera measurements made to simulate tropospheric-stratospheric structure measurements show the importance of careful control of focal plane noise and pattern noise. Focal plane cooling will certainly be required for visible, UV, and MWIR atmospheric structure sounding using the techniques investigated in this project.
- The feasibility of using Fourier Transform spectrometers to detect airglow and (by inference) atmospheric backscatter structure was demonstrated by the Kestrel FTHSI instrument operated against an airglow background. Again, the importance of focal plane and instrument cooling to minimize background noise was demonstrated.
- Measurements of airglow structure from the Shuttle using the GLO instruments shows that the virtual triangulation technique is feasible. Instrument line-of-sight ambiguities and spacecraft attitude changes limited our ability to obtain detailed pixel by pixel correlation, but the overall data (integrated over the focal plane width) verifies that the virtual triangulation technique works.
- The results of this limited effort are promising but not definitive. Therefore, more detailed simulations followed by ground-based experimental measurements using the airglow layers as the signal source are recommended before proceeding to a formal instrument design phase.

## 7. REFERENCES

- 1 L. Broadfoot, B.R Sandel, D. Knecht, R. Viereck and E. Murad, "A Panchromatic Spectrograph with Supporting Monochromatic Imagers," *App. Opt.*, 31 (16), 3083-3096, 1992. and A. L. Broadfoot, "Remote Sensing from the Space Shuttle and Space Stations," *Adv. Space Res.* 19, 623-626, 1997.
- 2 L. A. Strugala, R. D. Sears, et. al. *Optical Engineering*, 32 ( 5), 992-11001 May 1993.
- 3 R. D. Sears, "Simulation of Low Altitude Structure Using Multi-Component Stochastic Perturbation Models," 33<sup>rd</sup> Aerospace Sciences Meeting, AIAA Paper No. 95-0058, 10 January, 1995.
- 4 E. E. Gossard, D. B. Sailors, V. R. Noonkester, *Guide to Computation and Use of Cross Spectra in Geophysics and Radio Physics*, NELC Technical Document 136, 22 July, 1971
- 5 see *Mathcad Users Guide*, versions 6 and 7, published by Mathsoft Inc, Cambridge MA.
- 6 J. W. Chamberlin, "Physics of the Aurora and Airglow," Academic Press, New York, 1961, pp 167-169.

## **8. APPENDIX 1 Kestrel Corporation Experiment Report**

This report summarizes the ground-based AIRCAM and FTHSI experiment operating parameters as conducted by the Kestrel Corporation.





6624 Gulton Court NE  
Albuquerque, NM 87109  
(505) 345-2327

## TEST PLANNING SHEET

PROJECT: Vanguard Phase 1 SBIR  
DATE: 15 September, 1998

TEST DESCRIPTION:  
AirCam BACKSCATTER MEASUREMENTS.

TEST LOCATION:  
Kestrel Corporation Facility parking lot.

TESTS TO BE CONDUCTED:

- Low Altitude Atmospheric Structure and Wind Experiments per BMDO-SBIRP1- 980508.doc/09/10/98 section 2.1.1

SPECIAL HANDLING REQUIREMENTS:

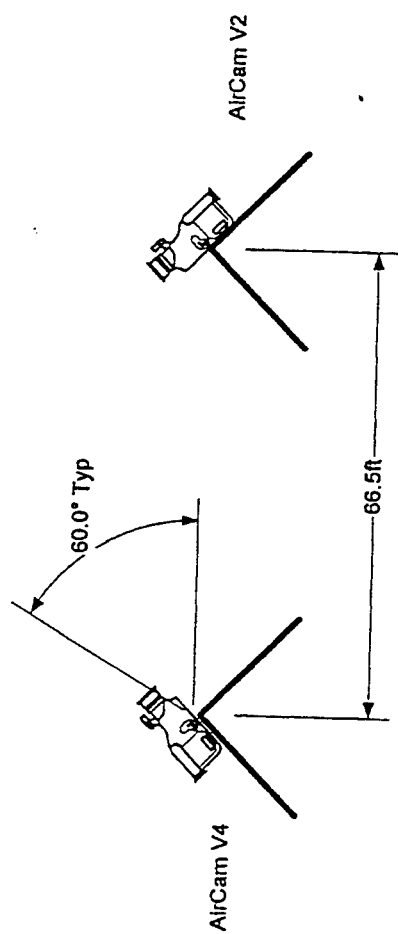
- Modify camera control software to allow for control of camera range and offset.

TESTING ASSIGNED TO: Al Jones DATE: 9/15/98

TESTING AUTHORIZED BY: John Otten DATE: 9/15/98

RESULTS RECIEVED BY: Robert D Sears DATE: 9/16/98

(PLEASE ATTACH ALL DOCUMENTATION TO THIS SHEET)



## AirCam Backscatter Experiment Separations

## BACKSCATTER TEST #2A

LOCATION:	KESTREL PARKING LOT			
TIME:	14:36			
DATE:	15-Sep-98			
ITEM	V4-B	V4-G	V2-B	V2-G
TYPE	9701	9701	9701	9701
LENS TYPE:	12-0302	12-0302	12-0302	12-0302
FILTER NO.:	4	1	4	1
POLARIZER:	NONE	NONE	NONE	NONE
DIFFUSER:	NONE	NONE	NONE	NONE
F-STOP:	1.4	2.4	1.4	2.4
GATING:	7	8	7	8
RANGE:	90	90	90	90
OFFSET:	60	60	60	60
ORIENTATION:	60 DeG	60 DeG	60 DeG	60 DeG
IMAGE #S (V2)	0-100			
IMAGE #S (V4)	101-200			

## BACKSCATTER TEST #2B

LOCATION:	KESTREL PARKING LOT			
TIME:	14:36			
DATE:	15-Sep-98			
ITEM	V4-B	V4-G	V2-B	V2-G
TYPE	9701	9701	9701	9701
LENS TYPE:	12-0302	12-0302	12-0302	12-0302
FILTER NO.:	1	4	1	4
POLARIZER:	VERTICAL	VERTICAL	VERTICAL	VERTICAL
DIFFUSER:	NONE	NONE	NONE	NONE
F-STOP:	1.4	2.4	1.4	2.4
GATING:	7	8	7	8
RANGE:	90	90	90	90
OFFSET:	60	60	60	60
ORIENTATION:	60 DeG	60 DeG	60 DeG	60 DeG
IMAGE #S (V2-B)	101-200	POLARIZER ON BLUE		
IMAGE #S (V4-B)	201-300	POLARIZER ON BLUE		
IMAGE #S (V2-G)	301-400	POLARIZER ON GREEN TIME 14:53		
IMAGE #S (V4-G)	401-500	POLARIZER ON GREEN TIME 14:53		

## Sheet1

## BACKSCATTER TEST #2C

LOCATION: KESTREL PARKING LOT

TIME: 14:47

DATE: 15-Sep-98

ITEM	V4-B	V4-G	V2-B	V2-G
TYPE	9701	9701	9701	9701
LENS TYPE:	12-0302	12-0302	12-0302	12-0302
FILTER NO.:	1	4	1	4
POLARIZER:	HORIZ	HORIZ	HORIZ	HORIZ
DIFFUSER:	NONE	NONE	NONE	NONE
F-STOP:	1.4	2.4	1.4	2.4
GATING:	7	8	7	8
RANGE:	90	90	90	90
OFFSET:	60	60	60	60
ORIENTATION:	60 DeG	60 DeG	60 DeG	60 DeG
IMAGE #S (V2-B)	201-300	POLARIZER ON BLUE		
IMAGE #S (V4-B)	301-400	POLARIZER ON BLUE		
IMAGE #S (V2-G)	401-500	POLARIZER ON GREEN TIME 15:09		
IMAGE #S (V4-G)	501-600	POLARIZER ON GREEN TIME 15:09		

## BACKSCATTER TEST #2D

LOCATION: KESTREL PARKING LOT

TIME: 15:15

DATE: 15-Sep-98

ITEM	V4-B	V4-G	V2-B	V2-G
TYPE	9701	9701	9701	9701
LENS TYPE:	12-0302	12-0302	12-0302	12-0302
FILTER NO.:	1	4	1	4
POLARIZER:	NONE	NONE	NONE	NONE
DIFFUSER:	YES	YES	YES	YES
F-STOP:	1.4	2.4	1.4	2.4
GATING:	7	8	7	8
RANGE:	90	90	90	90
OFFSET:	60	60	60	60
ORIENTATION:	60 DeG	60 DeG	60 DeG	60 DeG
IMAGE #S (V2)	501-530			
IMAGE #S (V4)	601-630			

## BACKSCATTER TEST #2E

LOCATION:	KESTREL PARKING LOT			
TIME:	15:19			
DATE:	15-Sep-98			
ITEM	V4-B	V4-G	V2-B	V2-G
TYPE	9701	9701	9701	9701
LENS TYPE:	12-0302	12-0302	12-0302	12-0302
FILTER NO.:	1	4	1	4
POLARIZER:	NONE	NONE	NONE	NONE
DIFFUSER:	COVER	COVER	COVER	COVER
F-STOP:	1.4	2.4	1.4	2.4
GATING:	7	8	7	8
RANGE:	90	90	90	90
OFFSET:	60	60	60	60
ORIENTATION:	60 DeG	60 DeG	60 DeG	60 DeG
IMAGE #S (V2)	531-560			
IMAGE #S (V4)	631-660			

## BACKSCATTER TEST #2F

LOCATION:	KESTREL PARKING LOT
TIME:	15:24
DATE:	15-Sep-98
ITEM	IR
TYPE	amber
LENS TYPE:	25mm
FILTER NO.:	3-5um
POLARIZER:	NONE
DIFFUSER:	NONE
F-STOP:	N/A
GATING:	AUTO
RANGE:	N/A
OFFSET:	AUTO
ORIENTATION:	ZENITH
IMAGE #S	1-100

## BACKSCATTER TEST #2G

LOCATION:	KESTREL PARKING LOT
TIME:	15:30
DATE:	15-Sep-98
ITEM	IR
TYPE	amber
LENS TYPE:	25mm
FILTER NO.:	3-5um
POLARIZER:	NONE
DIFFUSER:	WARM PLATE
F-STOP:	N/A
GATING:	AUTO
RANGE:	N/A
OFFSET:	AUTO
ORIENTATION:	ZENITH
IMAGE #S	101-130

## BACKSCATTER TEST #2H

LOCATION:	KESTREL PARKING LOT
TIME:	15:37
DATE:	15-Sep-98
ITEM	IR
TYPE	amber
LENS TYPE:	25mm
FILTER NO.:	3-5um
POLARIZER:	NONE
DIFFUSER:	COLD PLATE
F-STOP:	N/A
GATING:	AUTO
RANGE:	N/A
OFFSET:	AUTO
ORIENTATION:	ZENITH
IMAGE #S	131-160

9/15/1998	< Mission Flight Date	>
Van_Parkinglot1	< Mission Name <80 Char with Underscores	>
C:\VANPK1	< ACSYS Data File Storage Directory	>
4	< Red Camera Filter Reference Number	>
1	< Green Camera Filter Reference Number	>
4	< Blue Camera Filter Reference Number	>
1	< Yellow Camera Filter Reference Number	>
100	< Number of Images to be read Sequentially	>
35 46.00000	< Latitude Data Collection Point	>
-106 17.00000	< Longitude Data Collection Point	>
780-900	< Red Camera Filter 780,900	>
001024	< Red Camera Serial Number	>
24060149	< Red Camera Lens Serial Number	>
	< Blue Channel V4	>
412-525,455	< Green Camera Filter 412-525,hp-455	>
001022	< Green Camera Serial Number	>
24061149	< Green Camera Lens Serial Number	>
	< Green Channel V4	>
780-900	< Blue Camera Filter 780,900	>
001021	< Blue Camera Serial Number	>
24010078	< Blue Camera Lens Serial Number	>
	< Blue Channel V2	>
412-525,455	< Yellow Camera Filter 412-525,hp-455	>
002817	< Yellow Camera Serial Number	>
24009078	< Yellow Camera Lens Serial Number	>
	< Green Channel V2	>
3-5um	< IR Camera Filter 3-5um	>
0010	< IR Camera Serial Number	>
22240-084	< IR Camera Lens Serial Number	>

## **9. APPENDIX 2 University of Arizona Data**

This is the final report from Dr. Lyle Broadfoot, University of Arizona on GLO experiment data analysis.



## GLO 5 and GLO 6 Data

### Notes on Data in following pages

Page 1 – GLOVIEW – see description in “Home Page”

“[www.glo.lpl.arizona.edu/glo](http://www.glo.lpl.arizona.edu/glo)”

Page 2 – see notes on page

Files are named by the MET time of their recording

3053530A is 3/05:35:30 MET

The alpha character represents the CCD source

A, B, C, D, E are spectrographs

F, G, H are imagers

I is a tracking image

Other files are defined on the page

Page 3 - @ b.lst – a list of all records from the b-CCD, ie: b spectrograph

Page 5 – The Geometric Parameters in the header of each” \*.fit” files

Page 7 – A plot of the elevation history in data set 3d0535 and 3e0535

Data sets are named by the name of the first file in the data set

3d0535 means:

3<sup>rd</sup> day

d flight

05 hrs

35 min

GLO DATA file system:

<i>Flight</i>	<i>GLO#</i>	<i>DAY</i>	<i>Flight</i>	<i>hr</i>	<i>min</i>
STS-53	1	_____	_____	_____	_____
STS-63	2	_____	a	_____	_____
STS-69	3	_____	b	_____	_____
STS-74	4	_____	c	_____	_____
STS-85	5	_____	d	_____	_____
STS-85	5	_____	e	_____	_____

Page 8 – TH (tangent height) of GLO-5, GLO-6 in file 3d\_e0535.XLS.

Page 9 – Hyperspectral images of O<sub>2</sub>(0,0)

Page 10 – Spectra and Hyperspectral images

Page 13 – Plot of geometric data showing top and bottom of the slit (corrected Altitude)

Page 14 – Hyperspectral images of 5 emissions

<i>File name</i>	<i>GLO 6</i>	
60H5_1	6	OH(5,1)
60I6300	6	OI(6300)
6O2_0_0	6	O2(0,0)
6OI5577	6	OI(5577)
6Na5892	6	Na(5892)

Page 15 – as above for GLO 5

50H5_1	5	OH(5,1)
--------	---	---------

Page 16 – Constant altitude plots. The TH of 59 km was selected since that elevation was common to both instruments.  
Note the shift in maxima

Page 17 – 21

These plots show the relative position of the two traces @ 59kmTH for both instruments. The top plot shows the original position of the data in time. The bottom plot shows a shifted GLO-6 curve to match the GLO 5 curve. The time shift is noted.

They are in order of time delay between when the signature showed in the GLO-6 FOV pointing ahead and GLO-5 pointing behind shuttle track.

The difference in the latitude, longitude file was 6.7 min.

DATE/T 1-15-99 10:26:26

- o EC: 180 75 + RAM vector
- Sun • Jupiter • Moon

135



Earth

MET 03/05:59:00  
dMET 1.00 min

Hold: LULH  
Roll: 0.00  
Pitch: 180.00  
Yaw: 270.00  
GLO 5

90 +- Delay = 0.50

AZ: 64.97 EL: 22.16  
RA: 269.12 DEC: 8.74

SZA = 126

TNGT= 70,

LAT= -48,

LONG= 80,

Help Pause

Exit Clear

Box = TU

Stars = ( 1000.0) ALL3.STR  
1500, 1000, 500, 250, 0

GLO AZ/EL Grid

This is a reference list of the files in this Directory with some comments that apply to all data sets  
The File is dir.xls  
Directory of E:\3d0535

```

<DIR>      01-10-99  7:52a .
.. <DIR>    01-10-99  7:52a ..
3053530F FIT      17,280  1/10/99  7:45a  3053530F.FIT
3053530G FIT      17,280  1/10/99  7:45a  3053530G.FIT
3053530A FIT      63,360  1/10/99  7:45a  3053530A.FIT
3053530C FIT      63,360  1/10/99  7:45a  3053530C.FIT
3053530B FIT      63,360  1/10/99  7:45a  3053530B.FIT
3053530D FIT      63,360  1/10/99  7:45a  3053530D.FIT
3053530E FIT      63,360  1/10/99  7:45a  3053530E.FIT
3053556F FIT      17,280  1/10/99  7:45a  3053556F.FIT
3053556G FIT      17,280  1/10/99  7:45a  3053556G.FIT
3053556A FIT      63,360  1/10/99  7:45a  3053556A.FIT
3053556C FIT      63,360  1/10/99  7:45a  3053556C.FIT

3060603A FIT      63,360  1/10/99  7:47a  3060603A.FIT
3060603C FIT      63,360  1/10/99  7:47a  3060603C.FIT
3060603D FIT      63,360  1/10/99  7:47a  3060603D.FIT
3060603E FIT      63,360  1/10/99  7:47a  3060603E.FIT
3060618G FIT      17,280  1/10/99  7:47a  3060618G.FIT
3060618F FIT      17,280  1/10/99  7:47a  3060618F.FIT
3060618A FIT      63,360  1/10/99  7:47a  3060618A.FIT
3060618B FIT      63,360  1/10/99  7:47a  3060618B.FIT
3060618D FIT      63,360  1/10/99  7:47a  3060618D.FIT
3060618E FIT      63,360  1/10/99  7:47a  3060618E.FIT
3054306B         63,360  1/10/99  7:46a  3054306B

```

File Directory, All Files

864 file(s) 43,634,534 bytes  
2 dir(s) 509,362,176 bytes free

3D\_E0535 XLS 279,552 01-14-99 7:36a 3d\_e0535.xls

XL worksheets for geometric data

```

A      LST      7,145  1/12/99 10:19p  a.lst
B      LST      7,435  1/12/99 10:19p  b.lst
C      LST      7,087  1/12/99 10:19p  c.lst
D      LST      7,377  1/12/99 10:19p  d.lst
E      LST      7,493  1/12/99 10:19p  e.lst
F      LST      7,145  1/12/99 10:19p  f.lst
G      LST      6,333  1/12/99 10:19p  g.lst
H      LST      144    1/12/99 10:19p  h.lst
I      LST      144    1/12/99 10:19p  i.lst

```

Lists of data images from each detector in GLO instrument

```

UCAT  MAP      1,396  1/8/99  3:02p  ucat.map
GETMET OUT     27,464  1/8/99  1:35p  GETMET.OUT
MAKEHDF LST    23,083  1/11/99 3:55a  MAKEHDR.LST
DIR    TXT      0      1/14/99 7:54a  dir.txt
B      XLS     29,696  1/14/99 7:51a  b.xls

```

Ancillary data from JSC giving flight geometric information  
A file of MET times and exposure times prepared to request UCAT data  
Header of each spectral image in data set usually merged with UCAT data  
File DIR.  
b.lst in XL format

```

O2_0_1HY FIT    46,080  1/12/99 10:38a  O2_0_1HY.FIT
BG0_1HY FIT     46,080  1/12/99 10:38a  GBO_1.FIT
BG_REST FIT     46,080  1/12/99 10:38a  BG_REST.FIT
O2_0_0HY FIT    46,080  1/12/99 10:38a  O2_0_0HY.FIT
OH6_2HY FIT     46,080  1/12/99 10:38a  OH6_2HY.FIT
OH5_1HY FIT     46,080  1/12/99 10:38a  OH5_1HY.FIT
BG8_3HY FIT     46,080  1/12/99 10:38a  BG8_3HY.FIT
OH8_3HY FIT     46,080  1/12/99 10:38a  OH8_3HY.FIT
OI6300HY FIT    46,080  1/12/99 10:38a  OI6300HY.FIT

```

Hyperspectral Images

C. R. S.

Volume in drive E has no label  
Volume Serial Number is 4254-0301

# Directory of E:\3dQ535

3053530B FIT	63,360	1/10/99 7:45a	3053530B.
3053556B FIT	63,360	1/10/99 7:45a	3053556B.
3053567B FIT	63,360	1/10/99 7:45a	3053557B.
3053612B FIT	63,360	1/10/99 7:45a	3053612B.
3053626B FIT	63,360	1/10/99 7:45a	3053626B.
3053641B FIT	63,360	1/10/99 7:45a	3053641B.
3053703B FIT	63,360	1/10/99 7:45a	3053703B.
3053711B FIT	63,360	1/10/99 7:45a	3053711B.
3053726B FIT	63,360	1/10/99 7:45a	3053726B.
3053741B FIT	63,360	1/10/99 7:45a	3053741B.
3053802B FIT	63,360	1/10/99 7:45a	3053802B.
3053810B FIT	63,360	1/10/99 7:45a	3053810B.
3053840B FIT	63,360	1/10/99 7:45a	3053840B.
3053902B FIT	63,360	1/10/99 7:46a	3053902B.
3053909B FIT	63,360	1/10/99 7:46a	3053909B.
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3053939B FIT	63,360	1/10/99 7:46a	3053939B.
3054001B FIT	63,360	1/10/99 7:46a	3054001B.
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3054026B FIT	63,360	1/10/99 7:46a	3054026B.
3054028B FIT	63,360	1/10/99 7:46a	3054028B.
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3054123B FIT	63,360	1/10/99 7:46a	3054123B.
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3054207B FIT	63,360	1/10/99 7:46a	3054207B.
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125 file(s) 7,920,000 bytes

0 dir(s) 508,788,736 bytes free

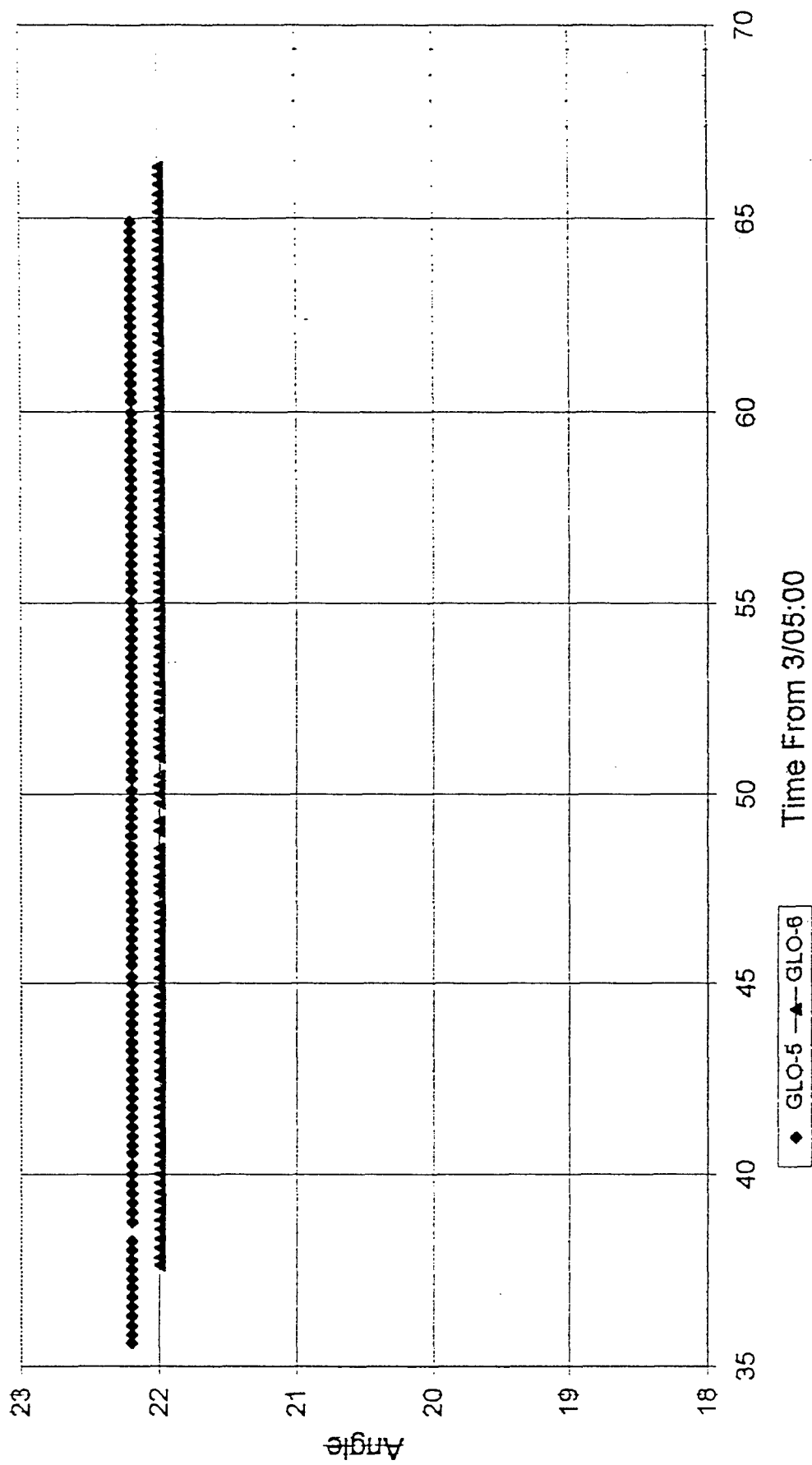
## Market List for 3059

52

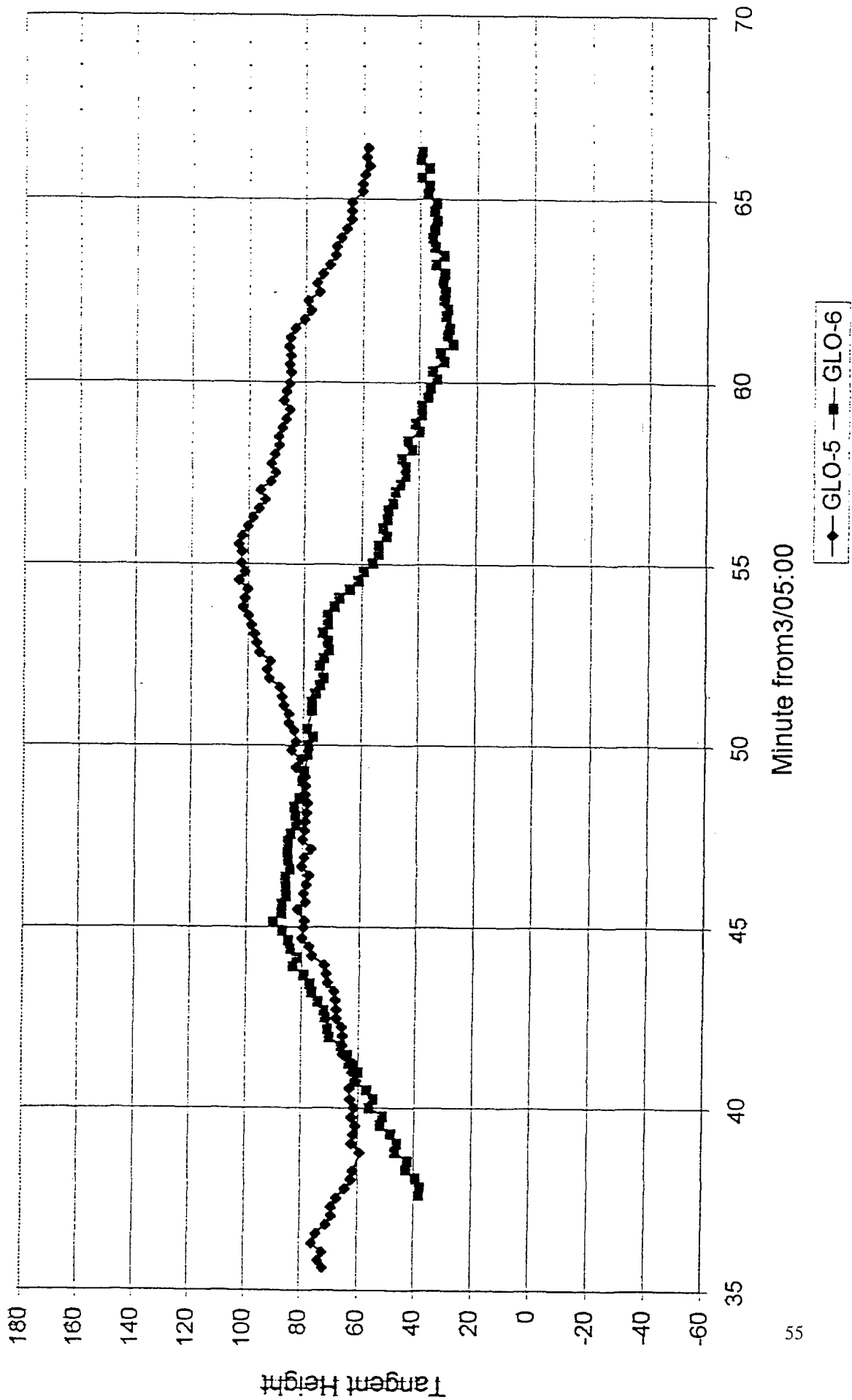


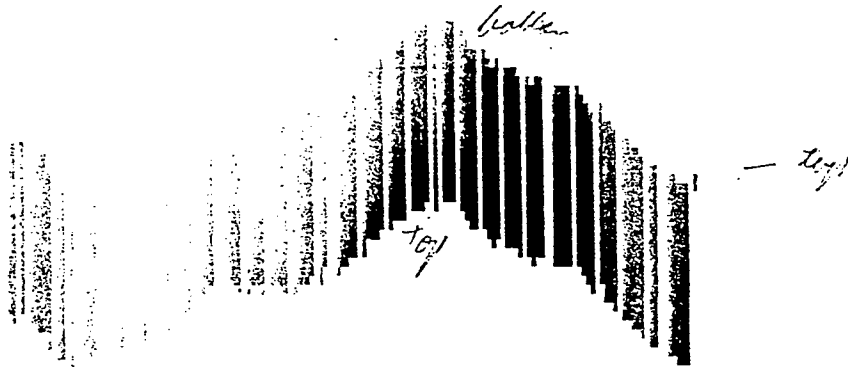


## 5 & 6 Elevation

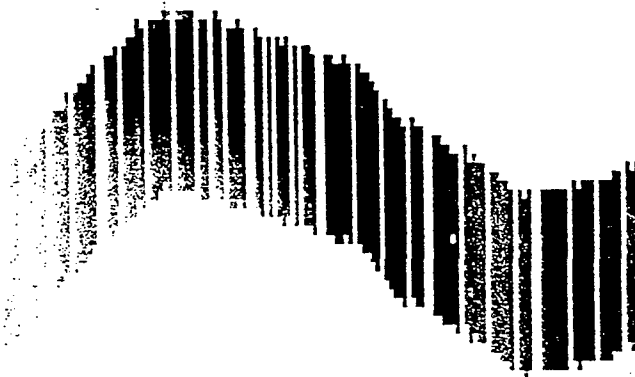


# Geometric Data; Set 3d\_e0535





I5 (1st -1) (3D0535): ( -10000)-( 90000)



I6 (1st -1) (3E0535): ( -10000)-( 90000)

12. 1000  
 12. 1000  
 12. 1000



I4 (1st -1) (3855488): ( 367)-( 786)

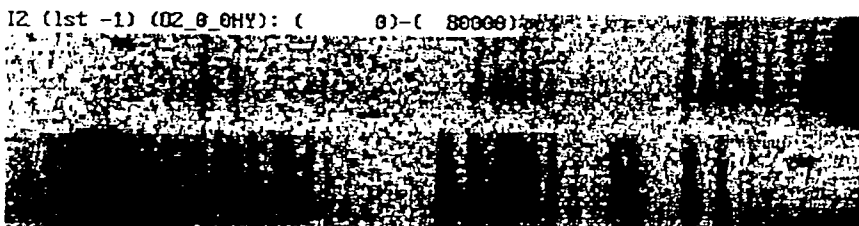
12. 1000



12. 1000

6. 4. 1000. 1. 1. 1.

12. 1000



I3 (1st 71) (38553588): ( 235)-( 388)

6. 4. 1000. 1. 1. 1.

5. 3. 1000. 1. 1. 1.

12. 1000  
 12. 1000  
 12. 1000



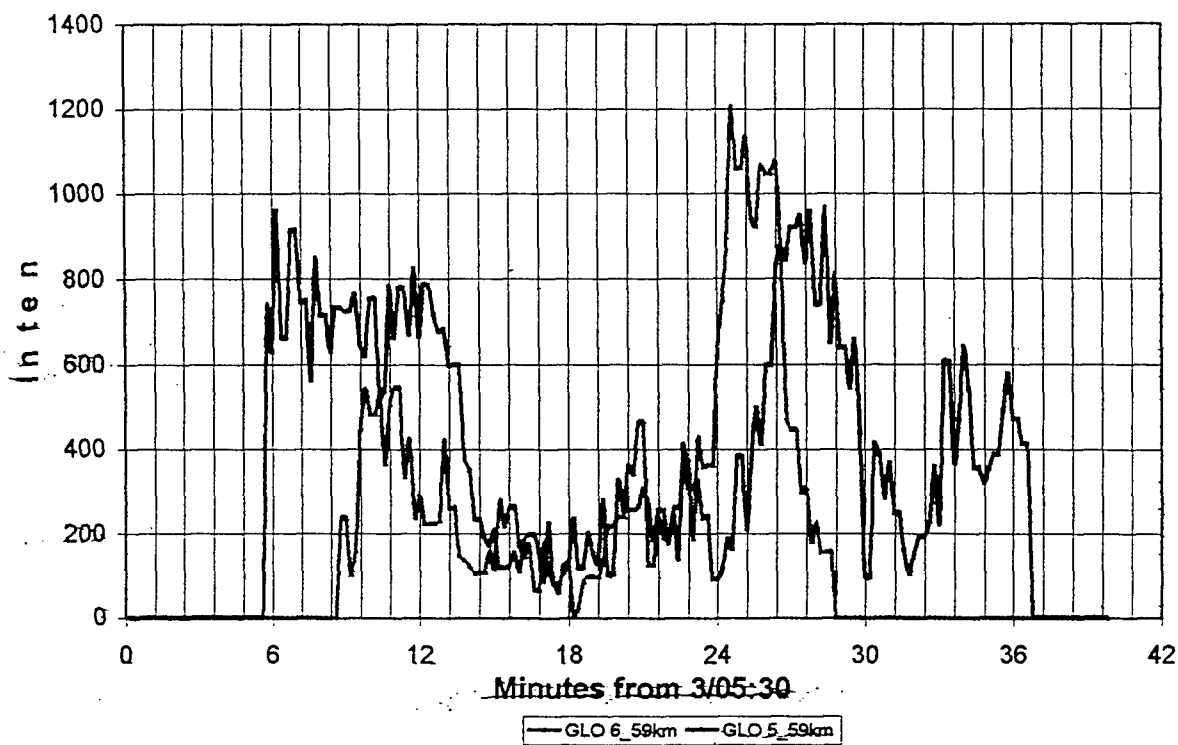
12. 1000

5. 4. 1000. 1. 1. 1.

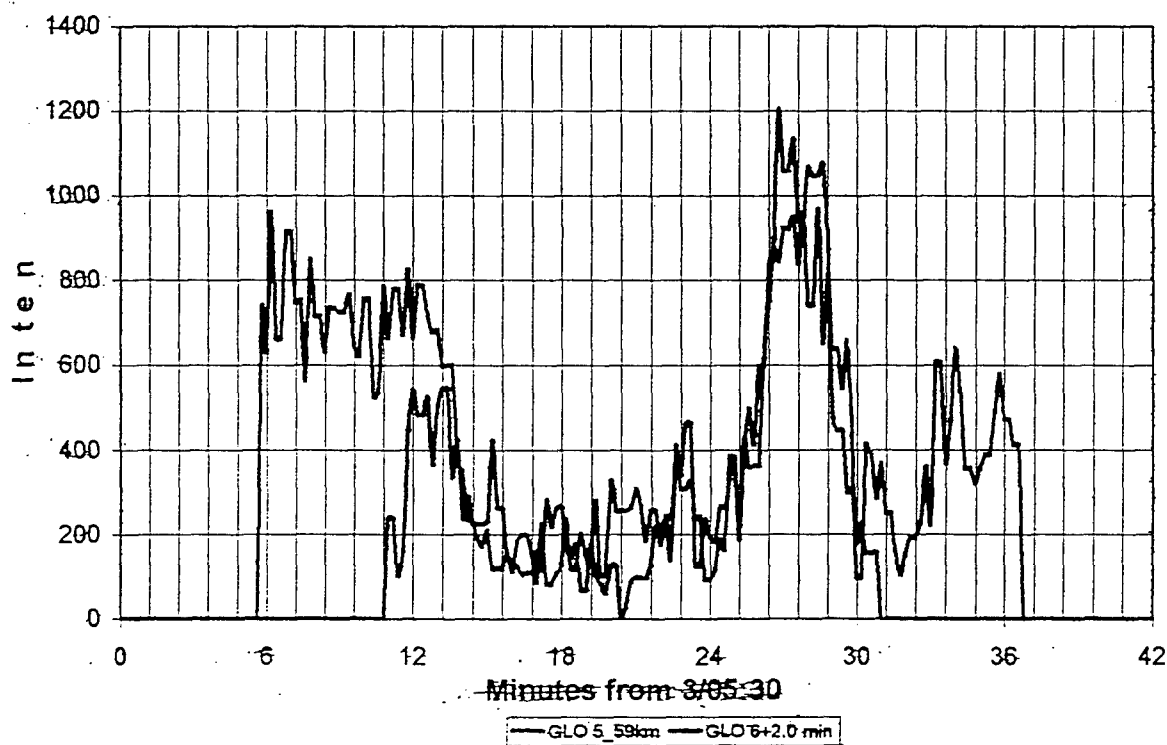
I1 (1st -1) (02\_0\_0HY): ( 0)-( 80000)

12. 1000 at 12. 1000 12. 1000 12. 1000

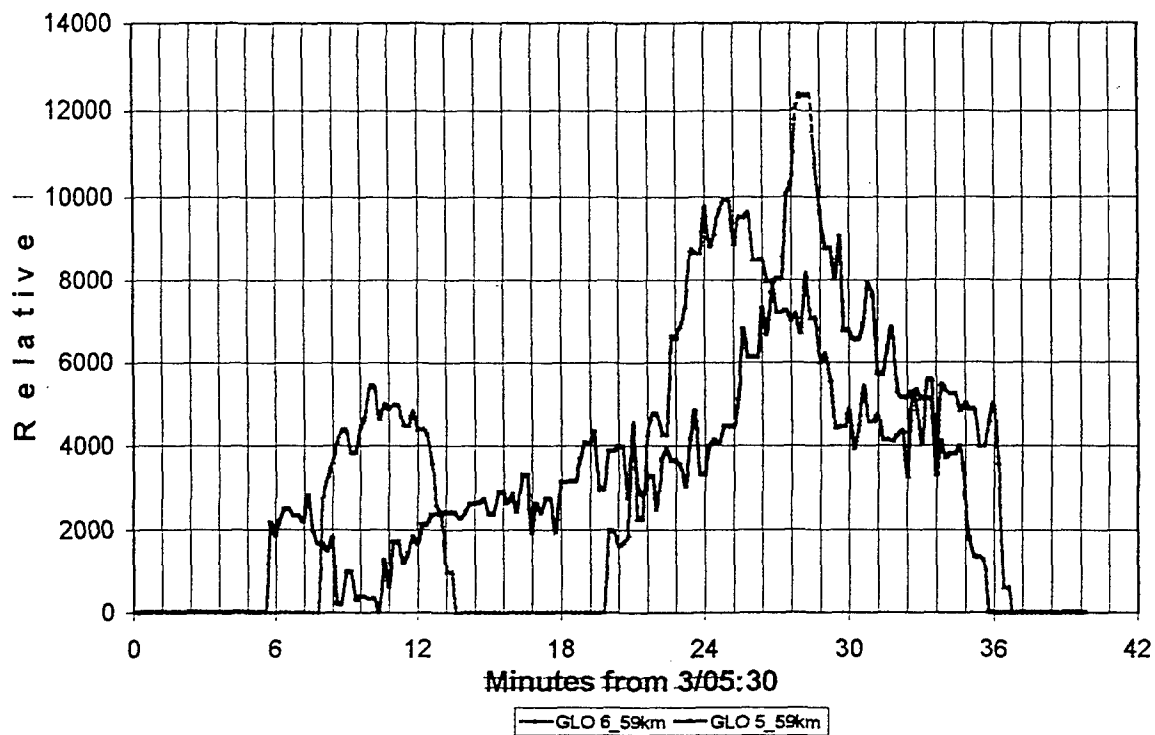
3e0535 OI(6300)



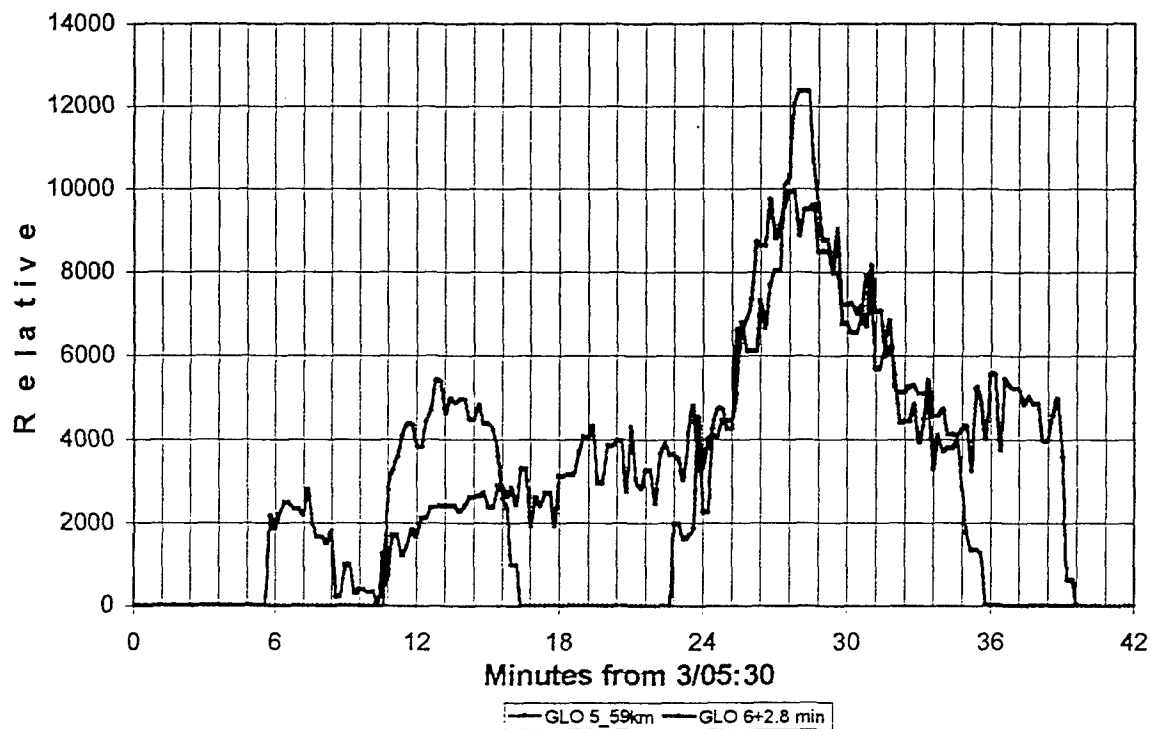
3e0535 OI(6300)



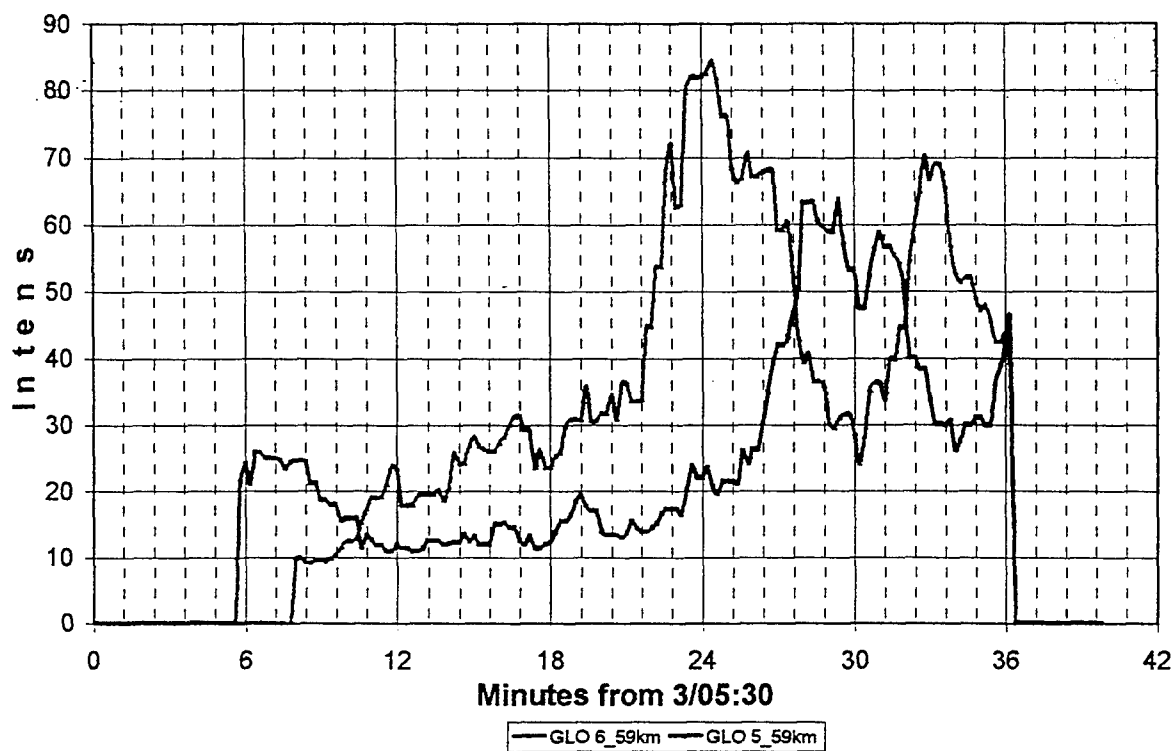
3e0535 OI(5577)



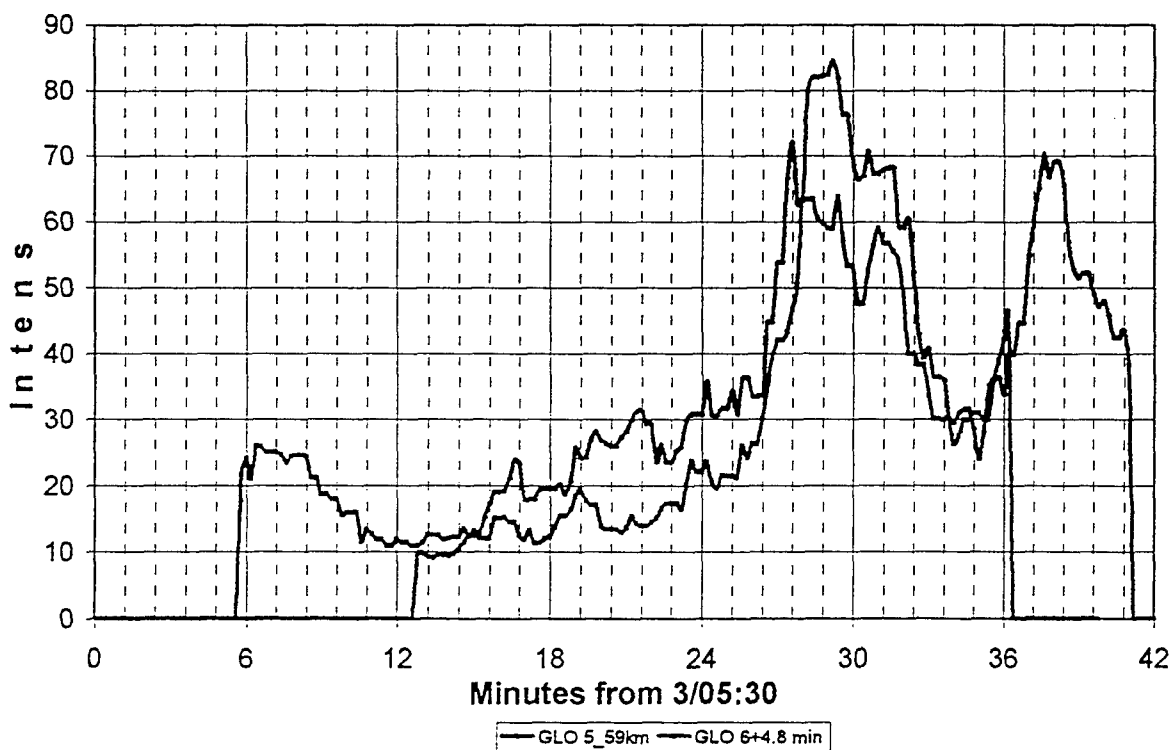
3e0535 OI(5577)



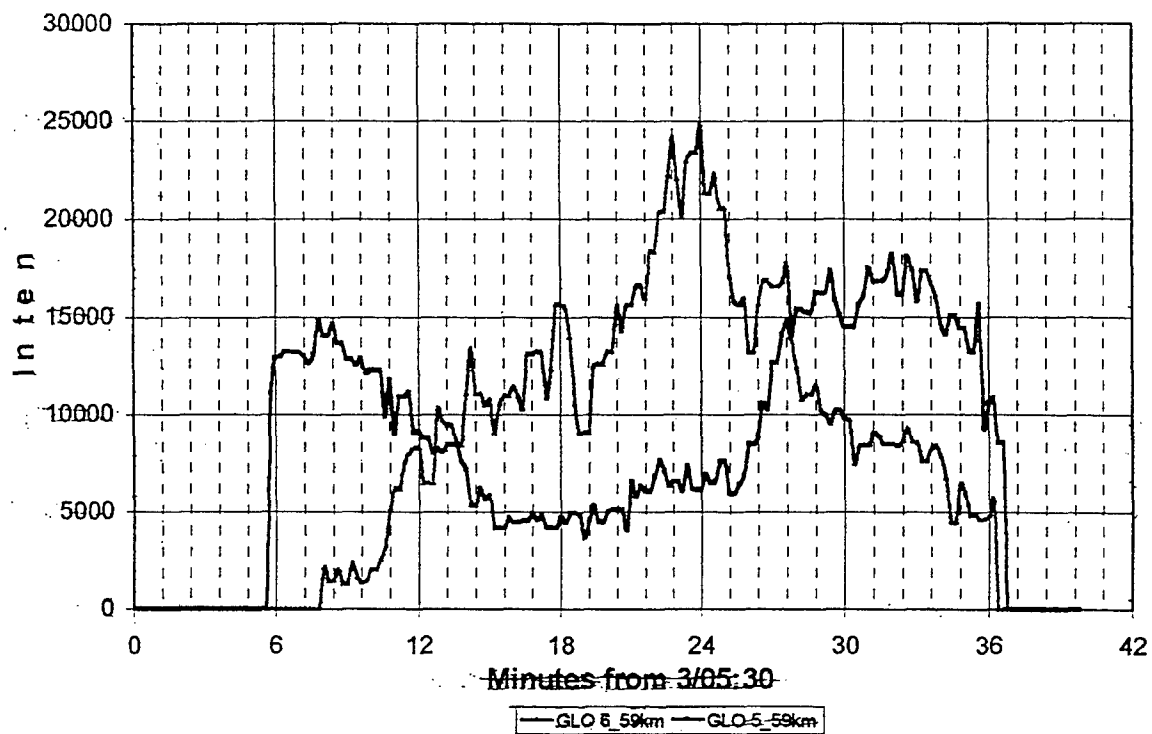
3e0535 O2(0,0)



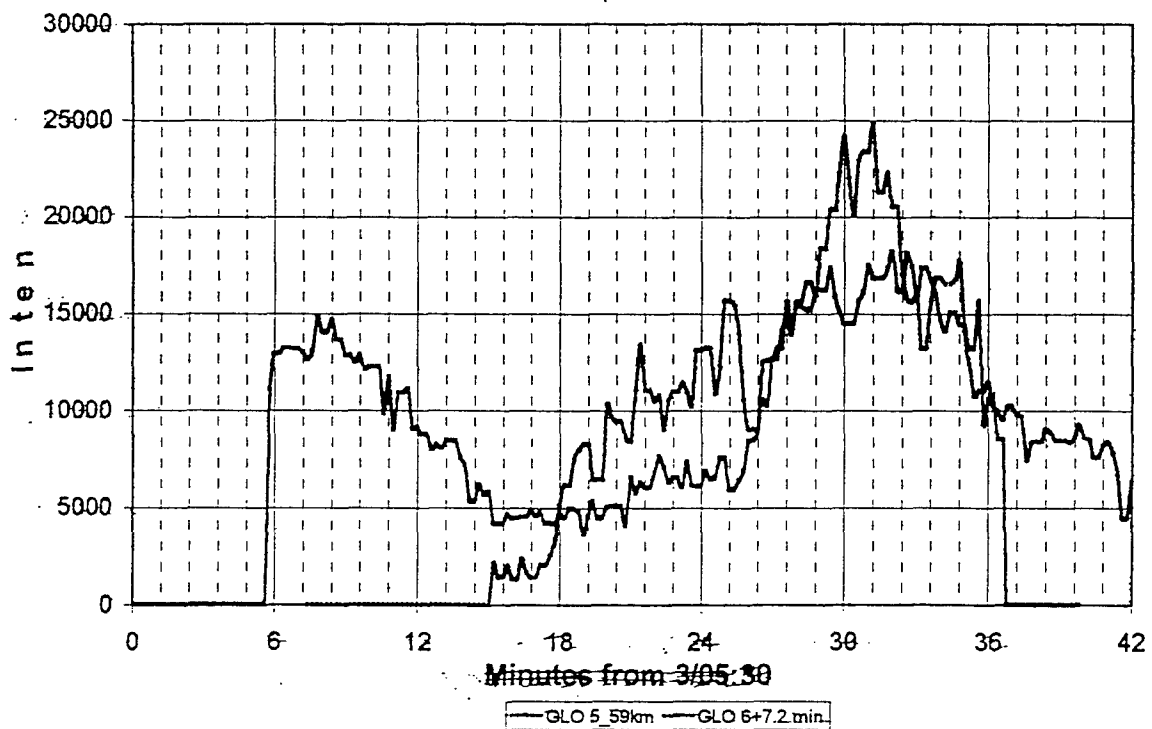
3e0535 O2(0,0)



3e0535 OH(5,1)



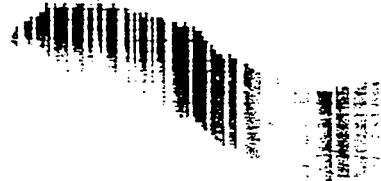
3e0535 OH(5,1)



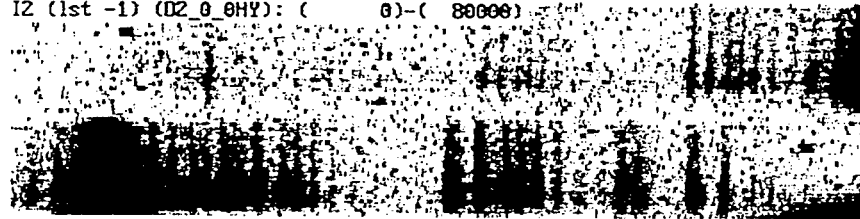




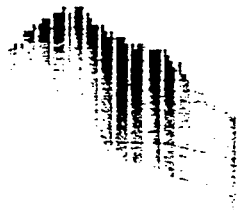
I4 (1st -1) (3055400B): ( 367)-( 786)



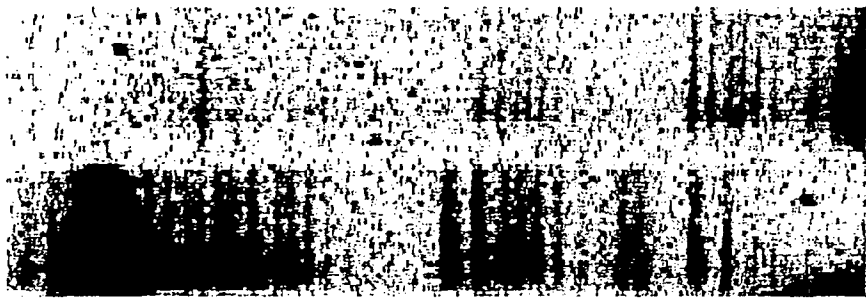
I2 (1st -1) (02\_0\_0HY): ( 0)-( 80000)



I3 (1st 71) (3055358B): ( 235)-( 389)

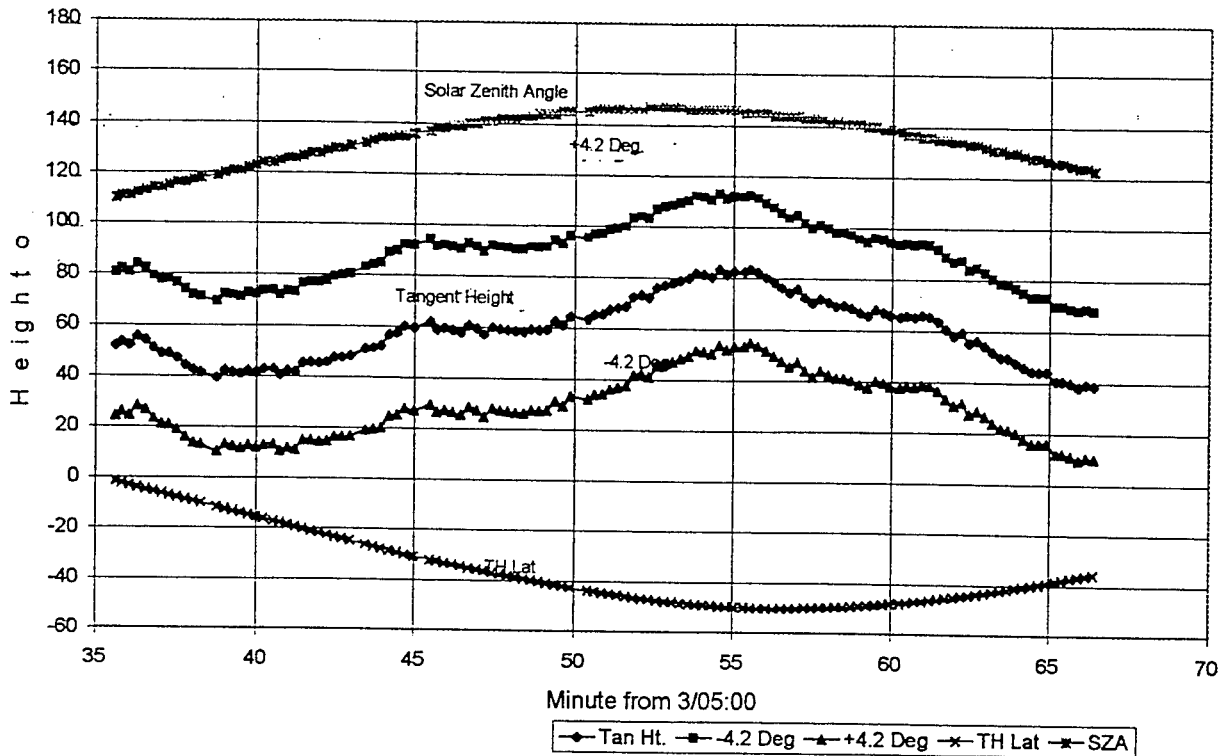


I1 (1st -1) (02\_0\_0HY): ( 0)-( 80000)

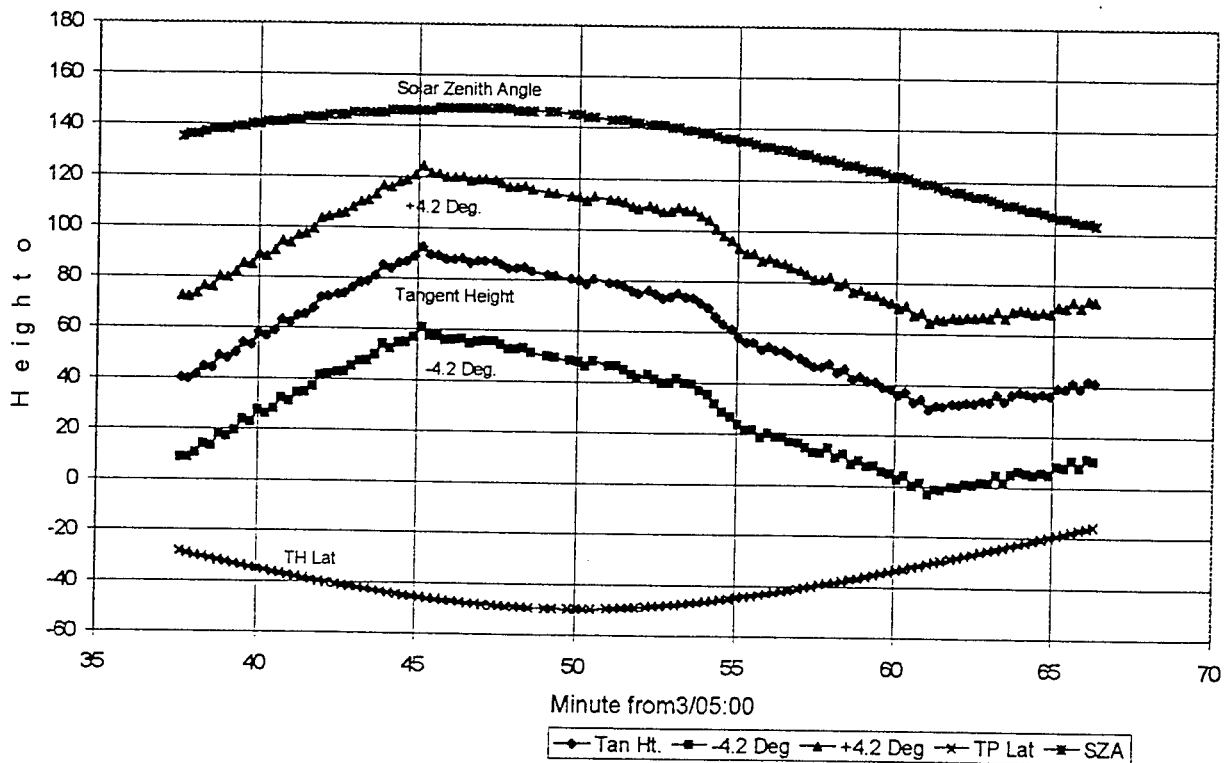


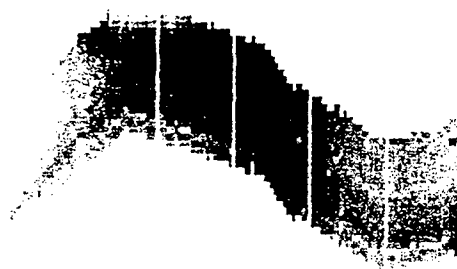
13 (1st 71) (30553588): ( 235)-( 300)

Geometric Data; Set 3d0535

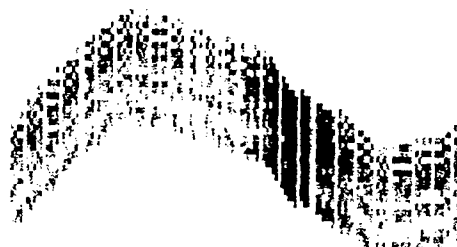


Geometric Data; Set 3e0535

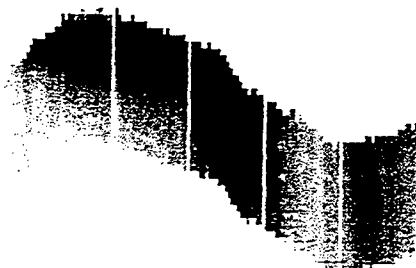




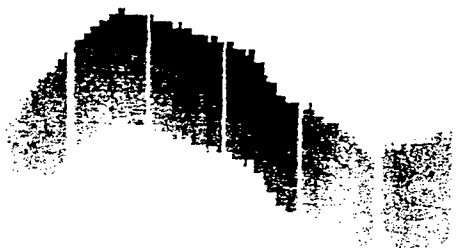
I6 (1st -1) (60H5\_1): ( 0)-( 28000)



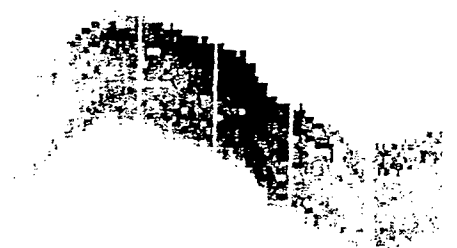
I5 (1st -1) (60I6300): ( 0)-( 1000)



I4 (1st -1) (602\_0\_0): ( 3000)-( 100000)



I3 (1st -1) (60I\_5577): ( 0)-( 15000)



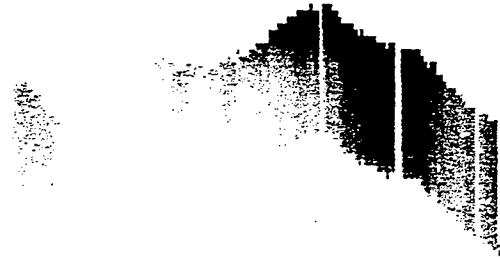
I4 (1st -1) (6MA\_5892): ( 0)-( 5000)



I3 (1st -1) (50H5\_1): ( 2000)-( 18000)



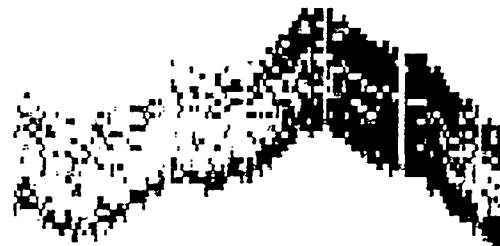
I2 (1st -1) (50I6300): ( 0)-( 900)



I1 (1st -1) (502\_0\_0): ( 1000)-( 90000)

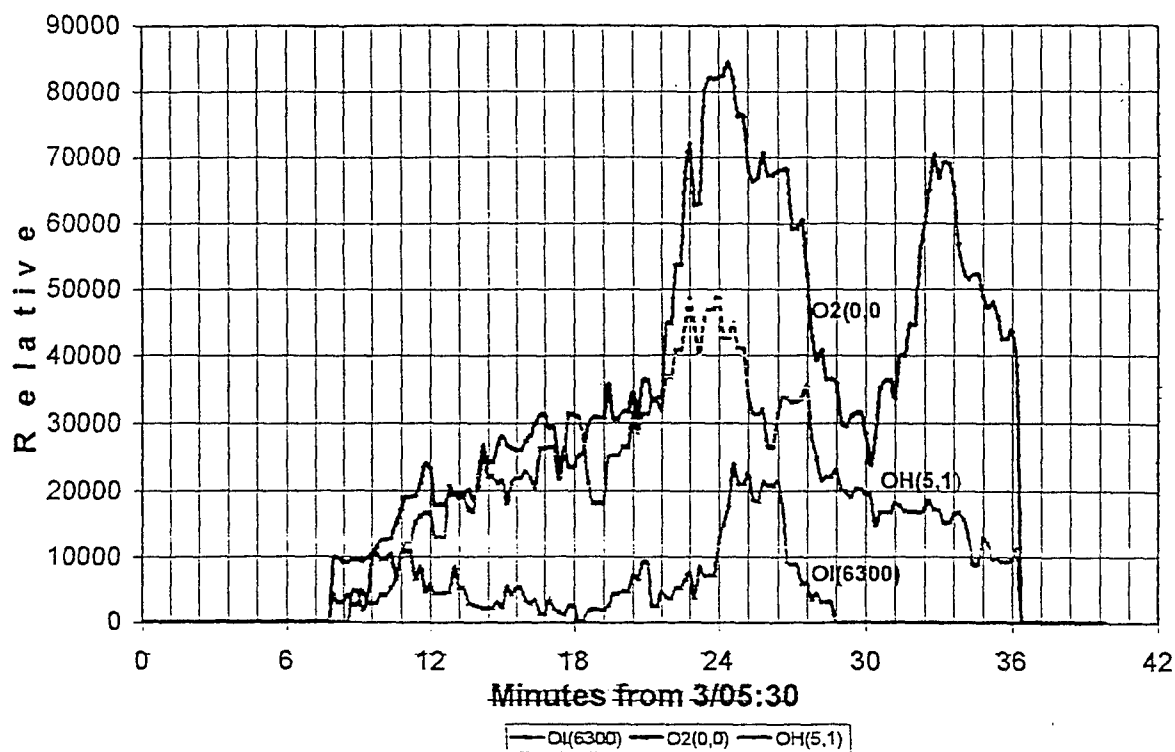


I1 (1st -1) (50I\_5577): ( 0)-( 9000)

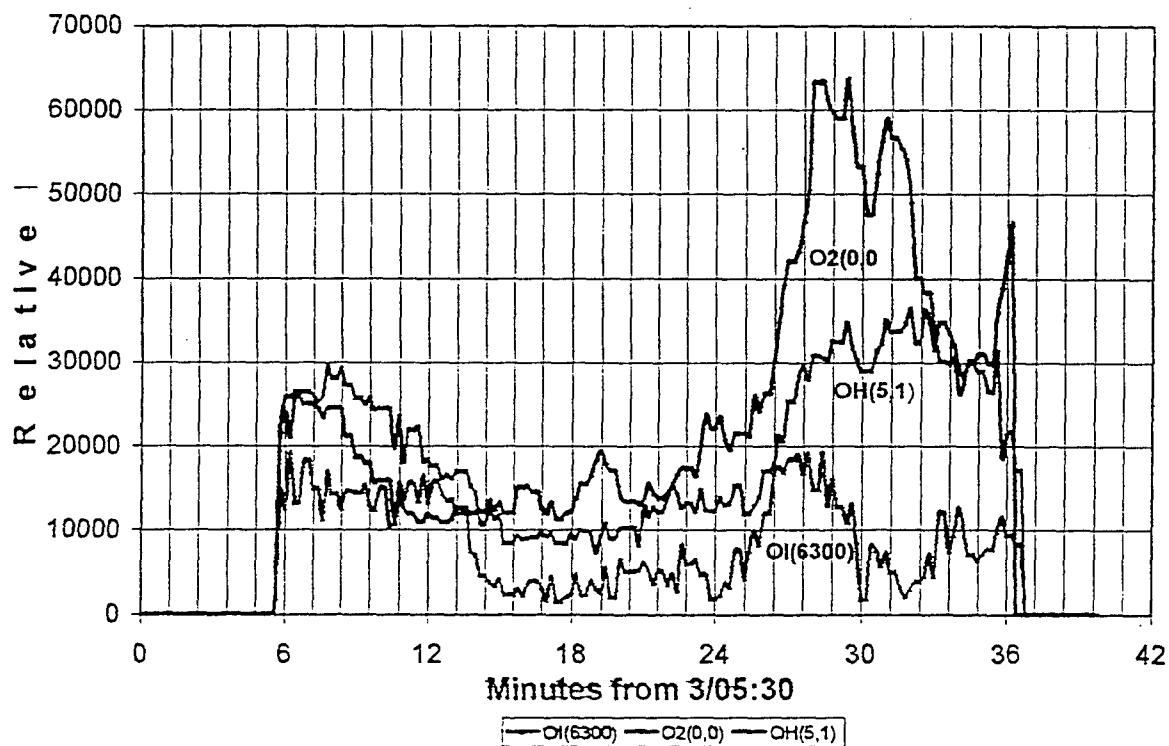


I2 (1st -1) (5MA\_5092): ( 500)-( 1000)

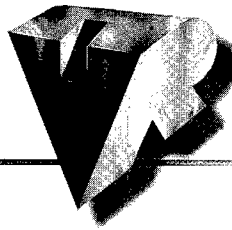
3e0535 GLO-6



3e0535 GLO-5



**Vanguard Research, Inc.**



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Fax (703) 273-9398

May 4, 2000  
00-0292

Defense Technical Information Center  
8725 John J. Kingman Road  
Suite 0944  
Ft. Belvoir, VA 22060-6218

Subject: Final Report – Replacement Page, SF298

Reference: Contract No. DASG60-98-M-0073

Dear Sir or Madam:

Please find enclosed two copies of a replacement page for SF298 for the final report. Please replace this page in your copies of the final report.

Should you have any questions, do not hesitate to contact me at (703) 934-6300.

Sincerely,

Debra A. Spear  
Contracts Administrator

Enclosure as stated

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